

# VORTEX

Vertically Optimized Research,  
Testing, & EXploration

## Manufacturing Status Review



Customer: Steve Borenstein  
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### Team

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# Project Overview



In order to expand the capabilities of the IRISS center and TORUS project in gathering meteorological data and understanding the formation of supercell thunderstorms, the VORTEX team will bring Vertical Takeoff and Landing (VTOL) functionality and extended endurance to the RiteWing Drak airframe.

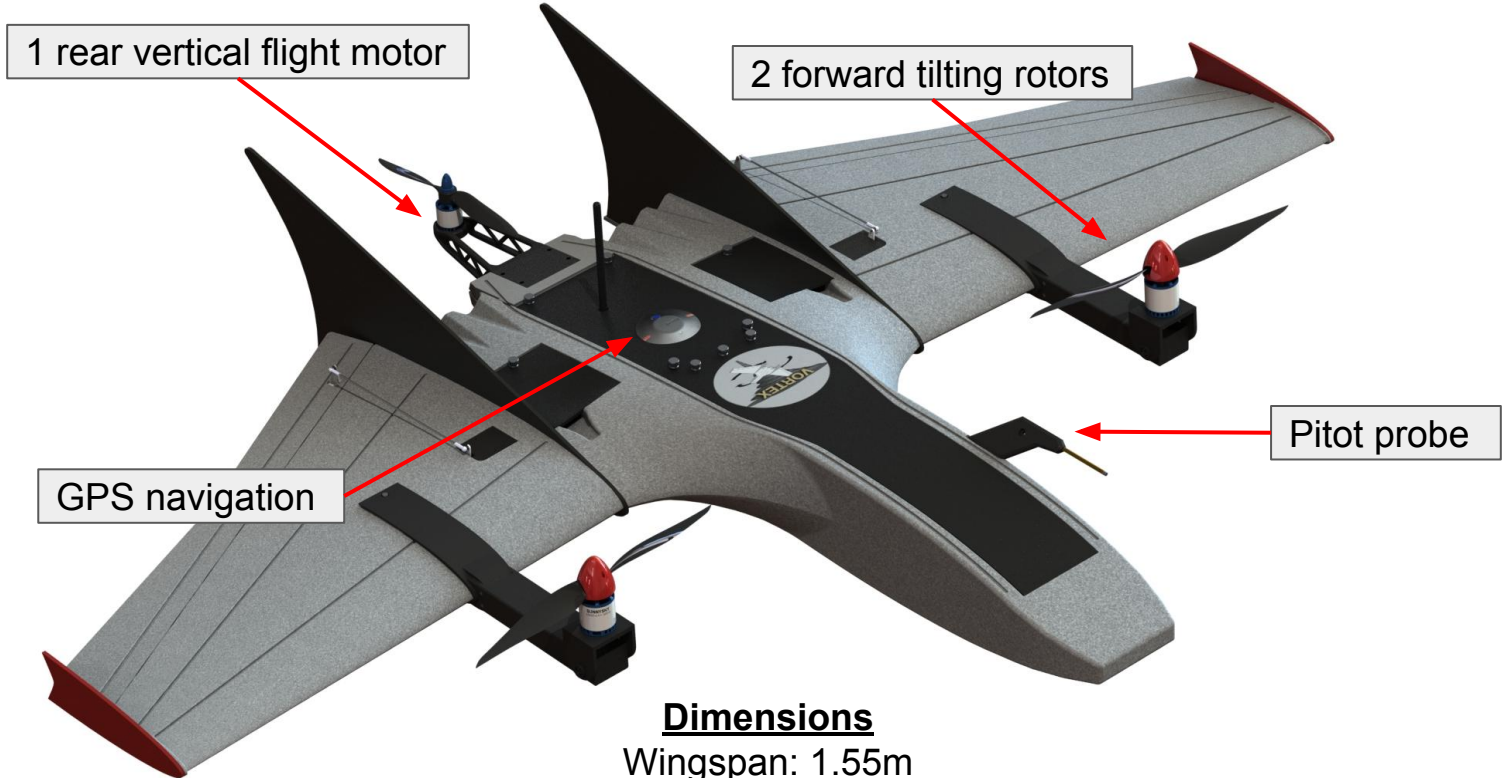


	Level 1	Level 2	Level 3
<b>Flight</b>	Show on a static test stand that the propulsion system is capable of producing enough thrust to provide a TWR greater than 1	Maintain tethered hover at 2m altitude for 30 seconds as well as demonstrate capability to transition to horizontal flight while aircraft is mounted to a test stand	Aircraft shall demonstrate takeoff ability via RAP Cat launch system as well as demonstrate full transition from vertical to horizontal flight modes.
<b>Budget</b>	The aircraft shall cost no more than \$1250, not including IRISS avionics package.	The aircraft shall cost no more than \$1000, not including IRISS avionics package.	The aircraft shall cost no more than \$900, not including IRISS avionics package.
<b>Endurance</b>	The propulsion system shall maintain required thrust output for the equivalent of 1 hour cruise and 2 takeoffs and landings (approximately 1 hr 16 minutes) on a static test stand in simulated freestream conditions of 18 m/s with >15% battery remaining	-	Demonstrate 1 hour of flight cruise as well as 2 takeoffs and landings
<b>Airframe</b>	A finite element analysis of the modified airframe will be performed to demonstrate that it can withstand the required forces with a FOS of 1.7	The aircraft will have full integration capabilities with RAPCat launch system, and show that it can withstand the forces due to acceleration.	The airframe shall with stand axial and lateral forces up to 10G.
<b>Avionics &amp; Electronics</b>	All motors and actuators shall be successfully integrated with the flight controller. The telemetry link shall be maintained with less than 25% packet loss within 1 km of the ground station.	All external (non-native) sensors are successfully integrated with the avionics system.	-

<b><u>Element</u></b>	<b><u>Justification</u></b>
Vertical Takeoff and Landing <b>(VTOL)</b>	Primary deliverable of project.
Structure <b>(STR)</b>	Structure must withstand forces of takeoff, flight, and landing.
Endurance <b>(END)</b>	Aircraft must be able to maintain flight for the required duration of 1 hour plus takeoffs and landings.
Automation <b>(AUT)</b>	Aircraft must autonomously perform mission flight profile as well as controlling takeoff, landing, and transitions.

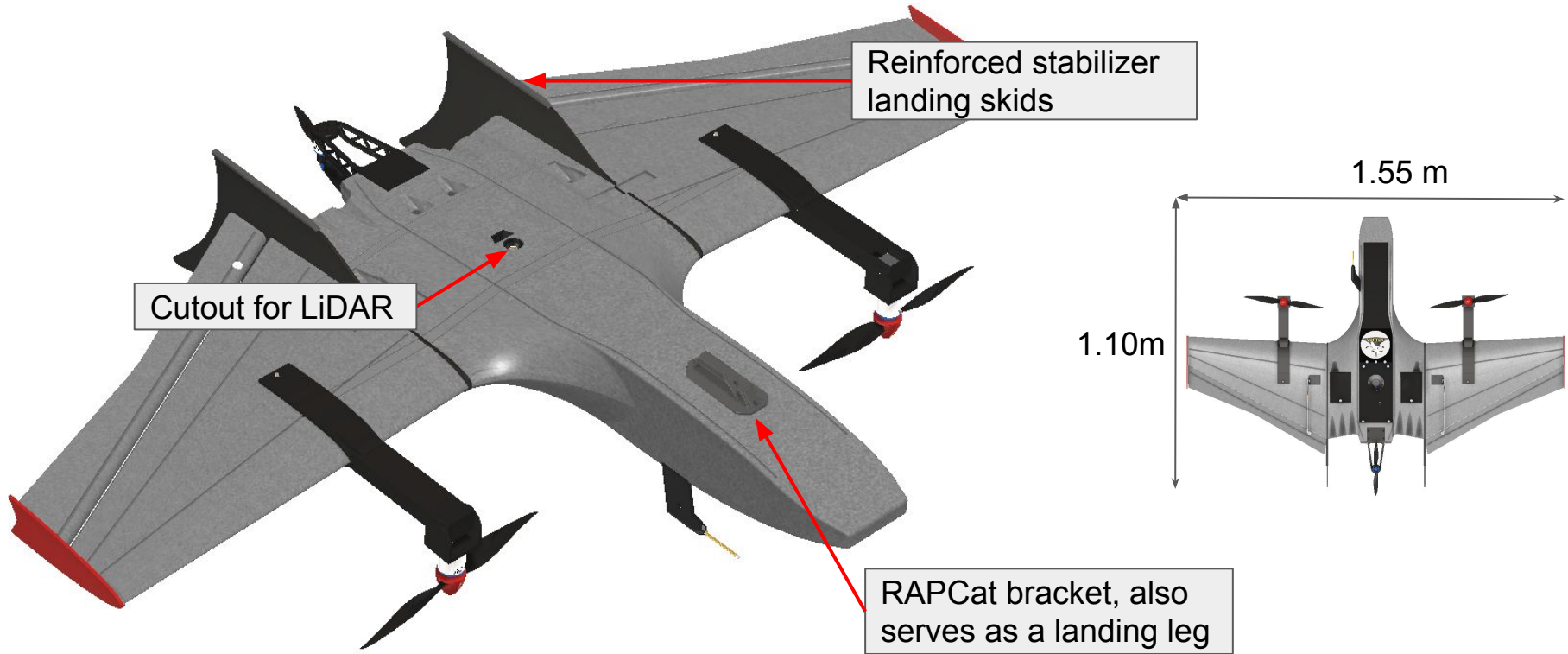


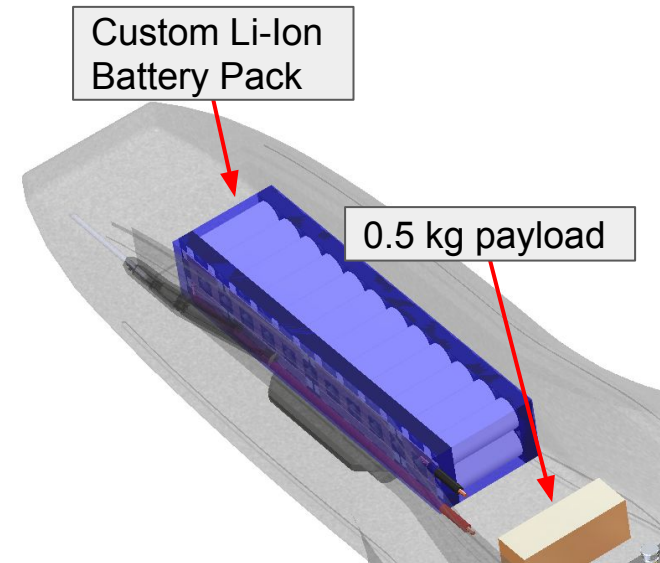
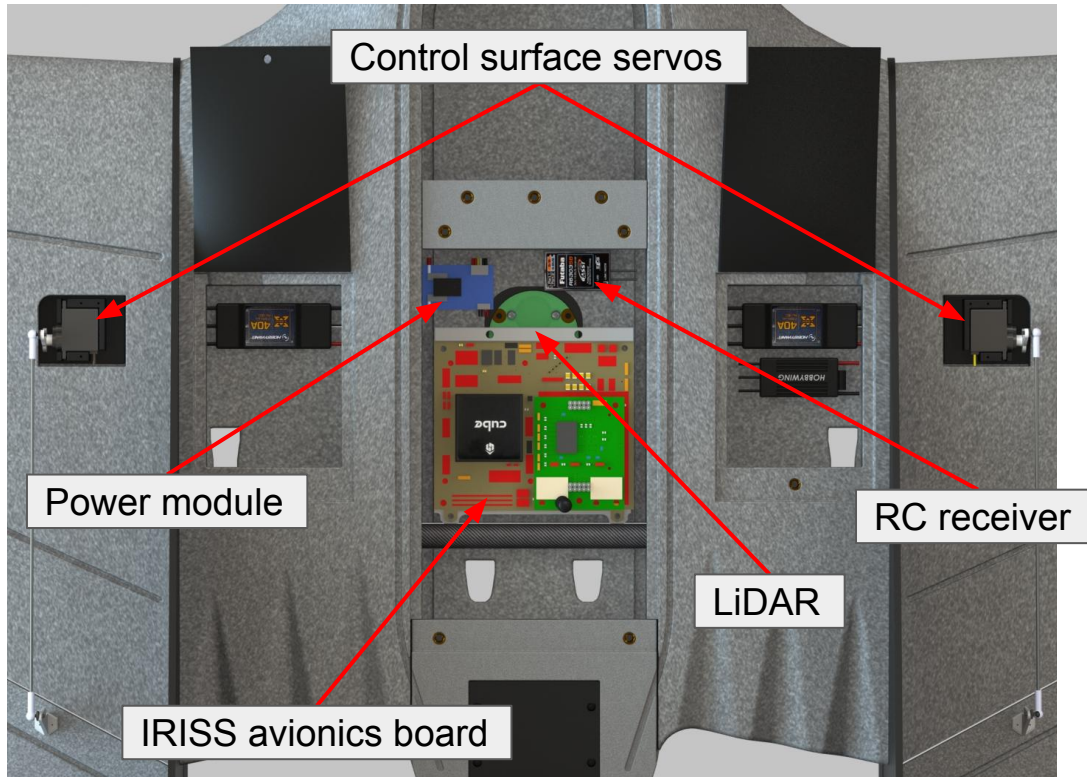
<b>FR1</b>	<b>VTOL</b>	The aircraft shall be a VTOL conversion of the COTS Ritewing RC “Drak” airplane kit
<b>FR2</b>	<b>END</b>	The aircraft shall have an endurance of 1 hour with 2 takeoffs and landings
<b>FR3</b>	<b>AUT</b>	The aircraft shall be able to autonomously execute all aspects of its mission from first takeoff through final landing
<b>FR4</b>	<b>AUT</b>	The aircraft shall maintain communication with the ground station up to a distance of 2km
<b>FR5</b>	<b>STR</b>	The aircraft shall be capable of carrying a 0.5kg payload
<b>FR6</b>	<b>STR</b>	The aircraft shall be capable of taking off from existing RAPCat launch system
<b>FR7</b>	<b>VTOL</b>	The airframe, propulsion system, and required mounting hardware shall cost no more than \$1000 per aircraft

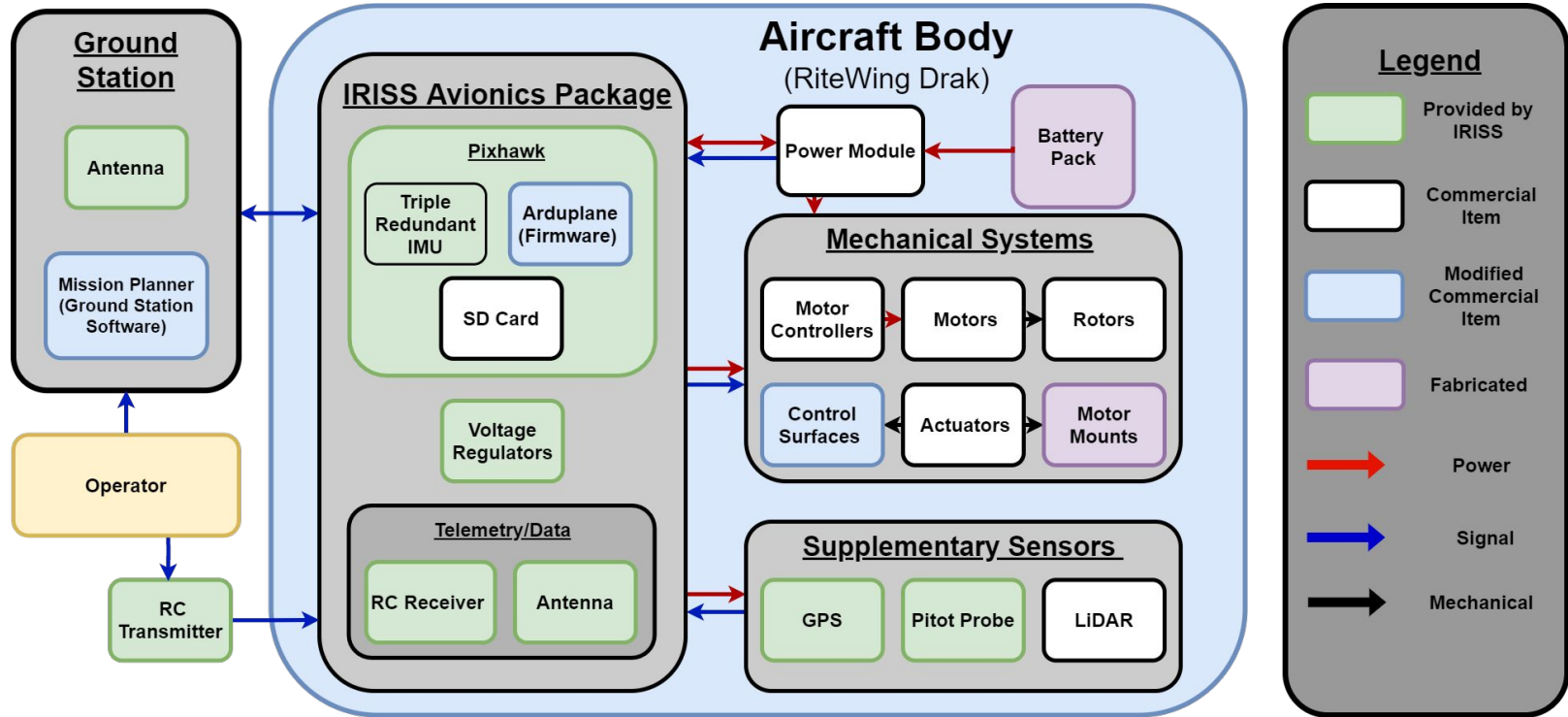


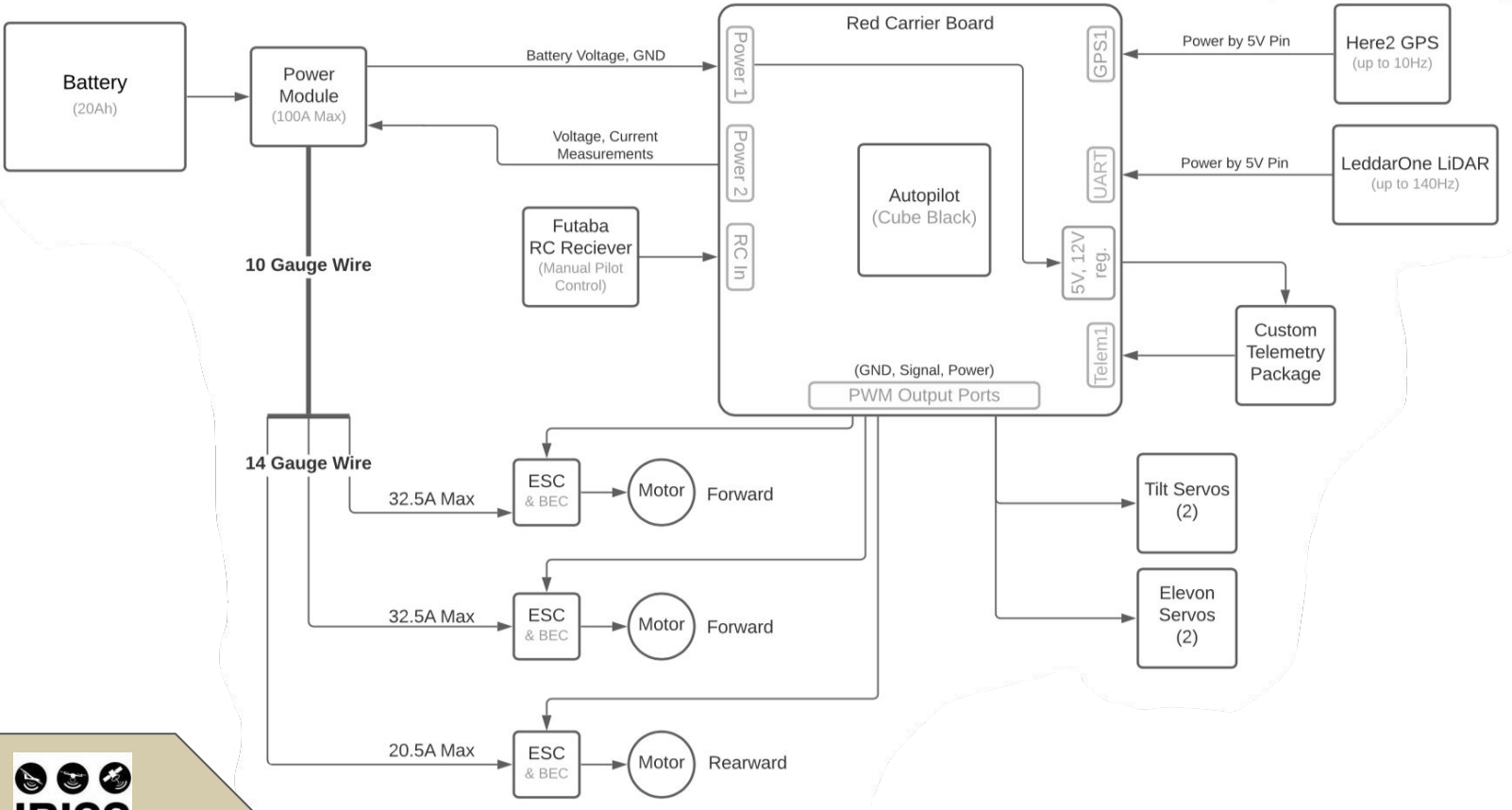
**Dimensions**  
Wingspan: 1.55m  
Nose-to-tail: 1.10m











# Changes Since CDR

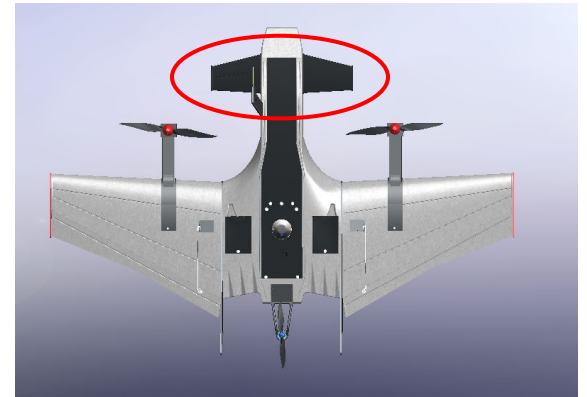
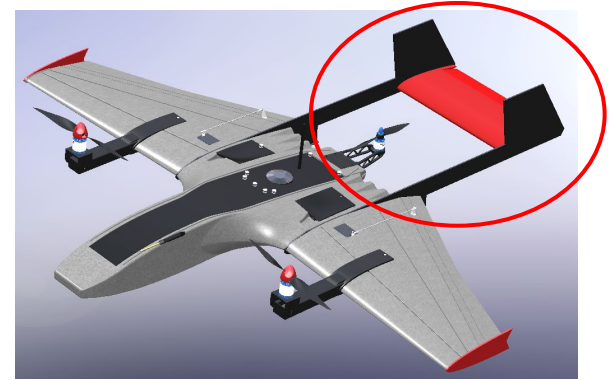
## Wing Motor Mount Testing

- Structural integrity concern due to custom wing motor mount
- Minor modifications to mounting bracket
- Additional testing to compare against FEM data
  - Assemble the component with a wing, load arm to 10G requirement, then load until failure



## Canard or Vertical Tail Addition

- Cruise conditions too close to stall
- Little room for maneuverability in operational flight
- New CFD iterations are too time consuming and have concerns with validation
  - Vehicle mounted test stand for obtaining aerodynamic data



## Static Equilibrium of Forces

$$\sum F = C_{L_c} q_\infty S_c + C_{L_{wb}} q_\infty S - W$$

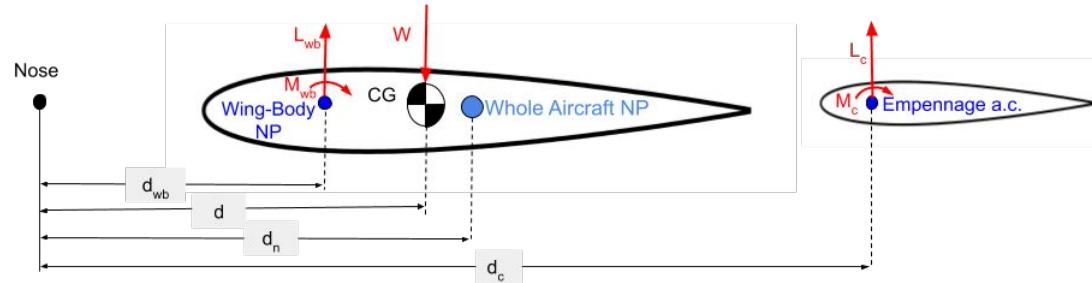
## Static Equilibrium of Moments

(about wing/body neutral point)

$$\sum M = C_{m_c} q_\infty S_c \bar{c}_c + C_{L_c} q_\infty S_c (d_{wb} - d_c) + C_{m_{wb_{a.c}}} q_\infty S \bar{c} - W (d_{wb} - d)$$

## Static Stability

$$K = \frac{1}{\bar{c}} \left[ \frac{C_{L_{c\alpha}} \frac{S_c}{S} d_c + C_{L_{wb\alpha}} d_{wb}}{C_{L_{c\alpha}} \frac{S_c}{S} + C_{L_{wb\alpha}}} \right]$$





- **LiDAR Verification**

- For verifying the accuracy of the LiDAR
- Tested by comparing the known heights of the stand to the height measured by the LiDAR

- **Dynamic Test**

- For measuring Lift, Drag, and Moment of the Aircraft
- Necessary because the Drak is too large for any wind tunnels we have access to
- Attach aircraft to test stand and drive at flight speed (~40 mph)
  - Test location approved at Colorado Air and Space Port - 10,000 ft of straight, private road

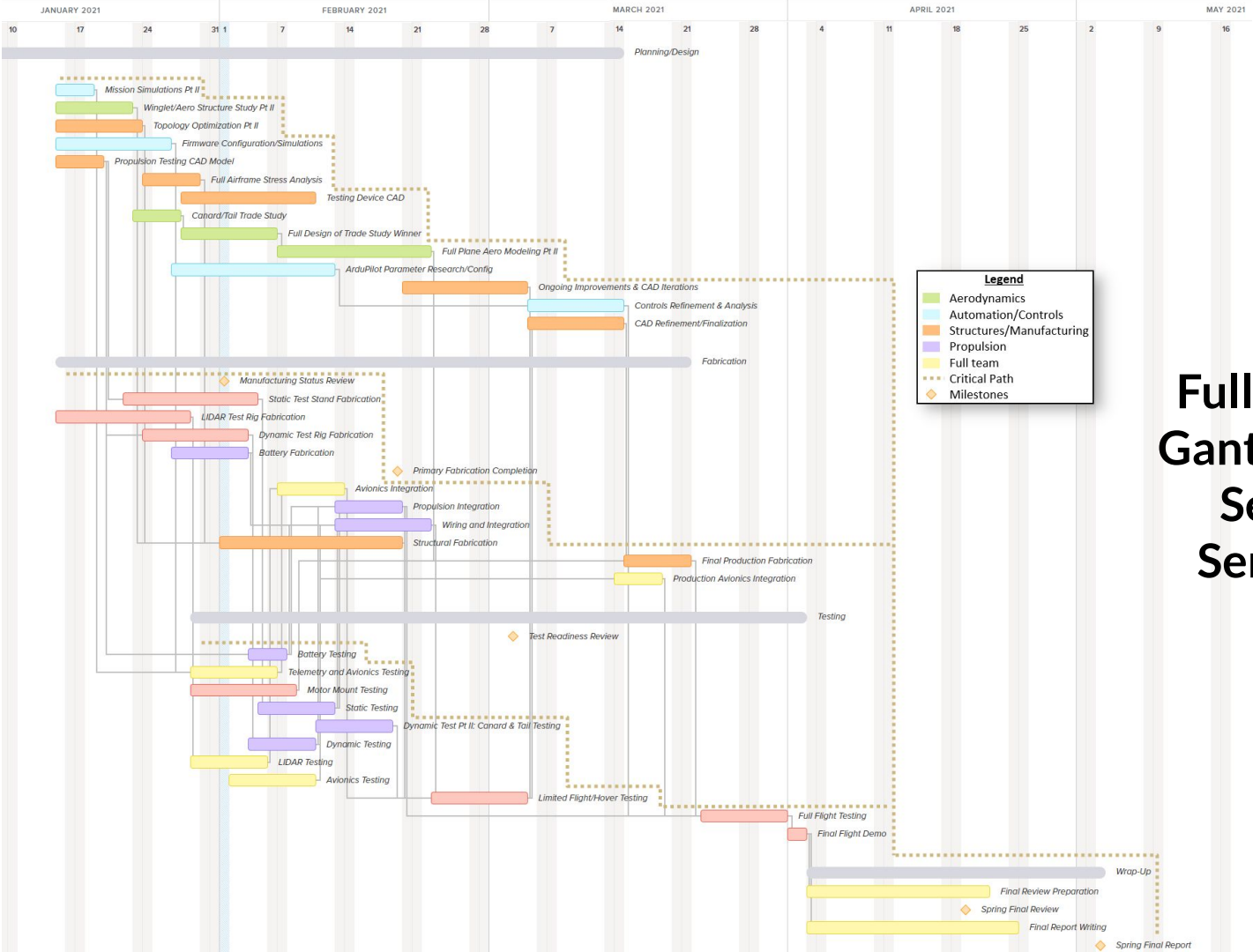
- **Static Test**

- For verifying manufacturer specifications of motor and propeller combinations
- DBF already has a stand but it requires safety improvements, entire test stand will be donated to DBF after project

- **Motor Mount Testing**

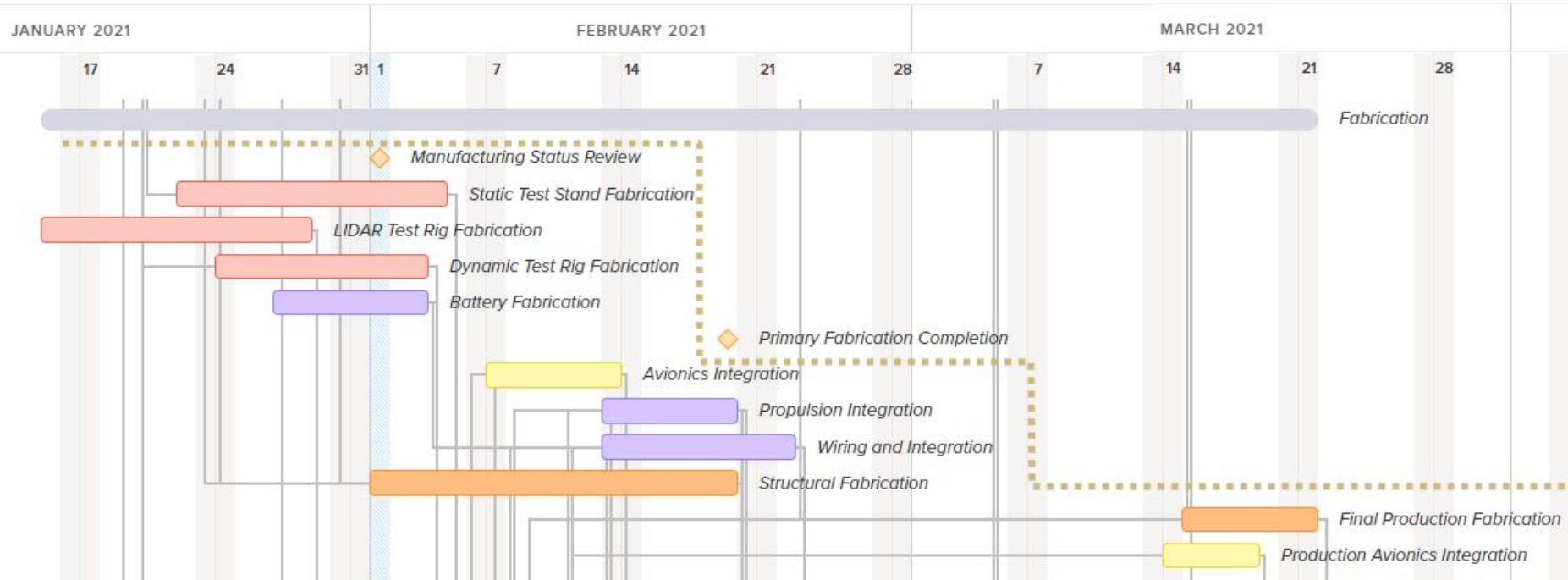
- Verify SolidWorks FEM
- Print and mount motor arm
- Apply expected maximum flight loading and measure deflection with calibrated camera
- Check foam for deformation or damage
- Apply loading to failure, record maximum loading values and behavior

# Schedule



# Full Project Gantt Chart - Second Semester

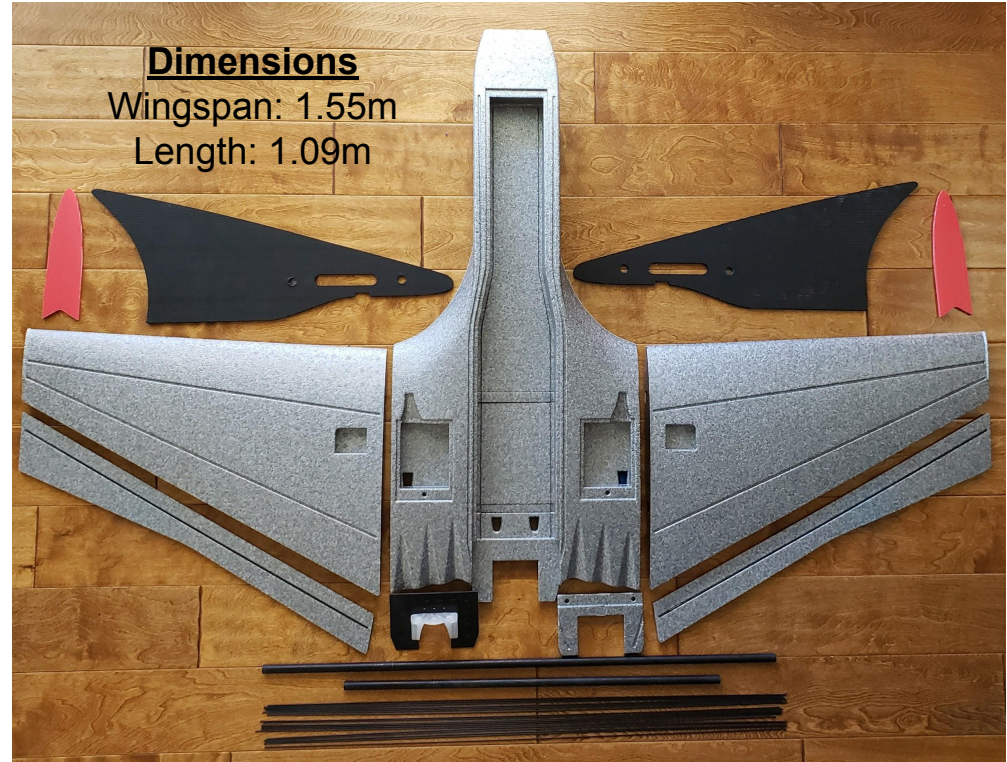
# Fabrication and Testing Gantt Chart



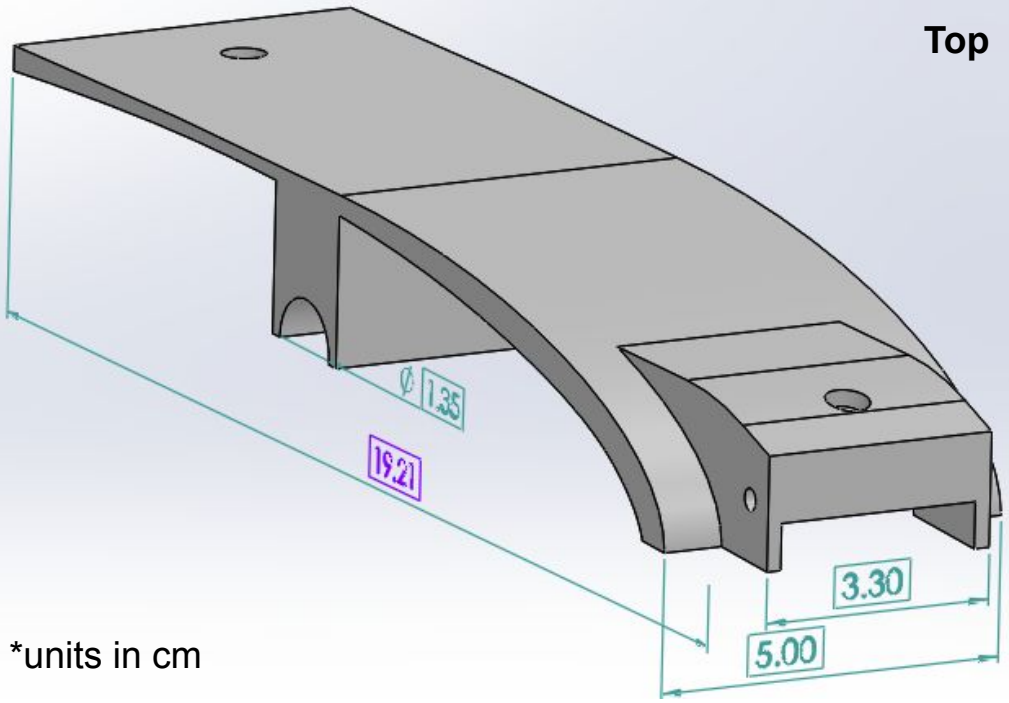
# Manufacturing

Component	Purchased or Manufactured	Manufacture Time	Completion Date	Status
Drak Wing Kit	Purchased/Assembled	~ 3 days	Feb 5th-8th	Waiting on Adhesive
Wing & Motor Mounts	Manufactured/Printed	12 hours	Jan 30th	Printed
Lidar Test Stand	Purchased/Printed	1 hour	Feb 5th	Printed, Waiting on Delivery
Custom Battery	Manufactured	6 hours	Feb 3rd	Waiting on R2R
Static Test Stand	Manufactured	3 hours	Feb 4th	Waiting on R2R
Dynamic Test Stand	Manufactured	4 hours	Feb 3rd	Waiting on Delivery

- Three Drak kits purchased and in our possession
  - Backups in case of damage during testing
- Extra wing set
  - Test wing mount loading
- Kit includes EPP foam body, coro-plast stabilizers, carbon fiber rods and spars
- Additional assembly tools
  - Adhesive
  - Tape for control surfaces



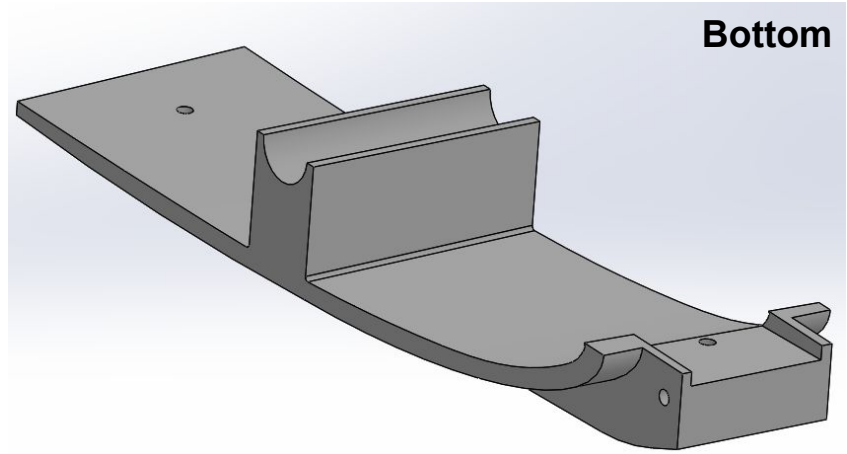
All Drak kits procured, waiting on adhesive to be delivered  
To be assembled by Feb 5th-8th



\*units in cm

- Wing mounts printed out of PETG
  - ~12 hours total
- Wing mounts held together by glue and forward & rear 3mm bolts in the wing

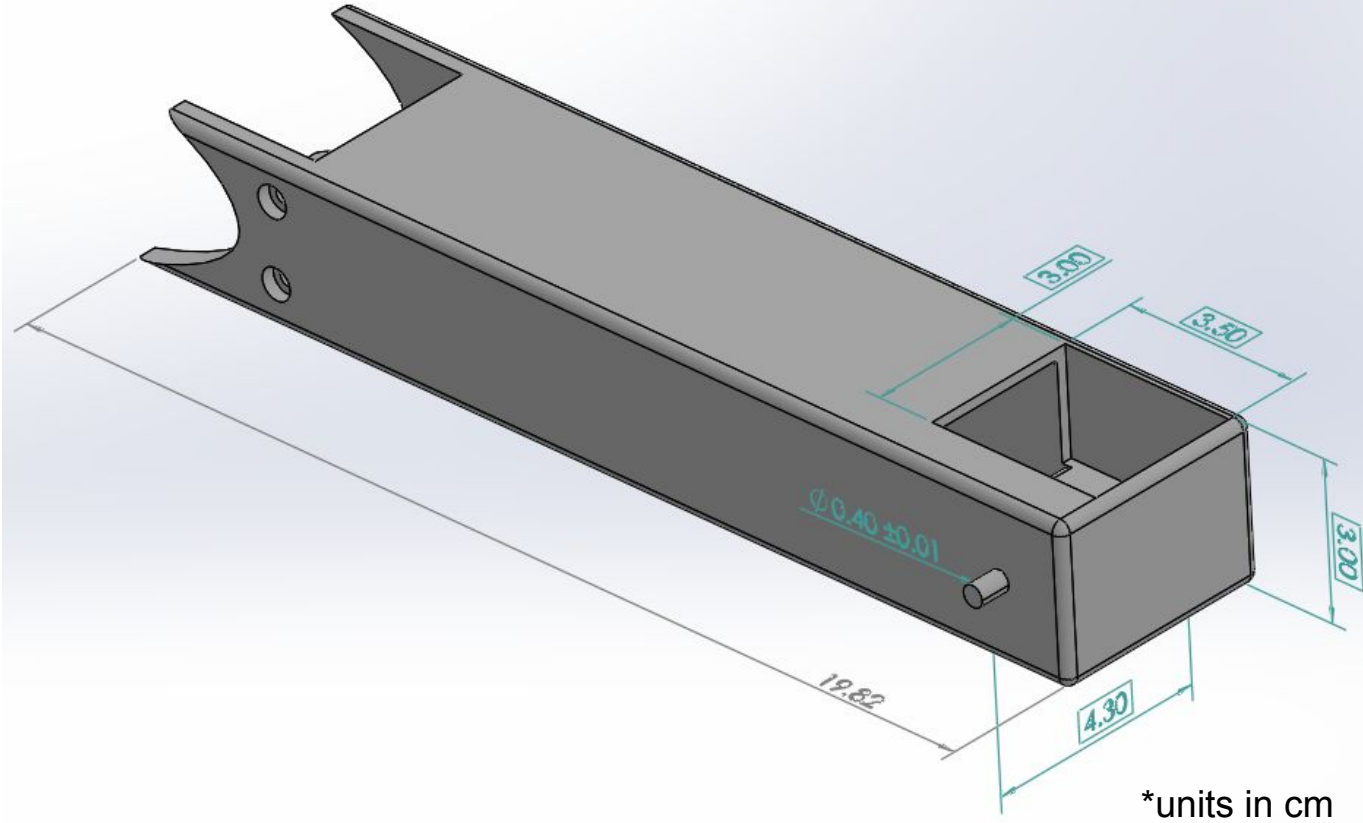
One set is printed and ready to begin wing load testing





- Arm mounts printed out of PETG
  - ~10 hours each
  - Hollow to allow for wire channeling
- Motor arm attaches to wing mount by two horizontal 3mm bolts

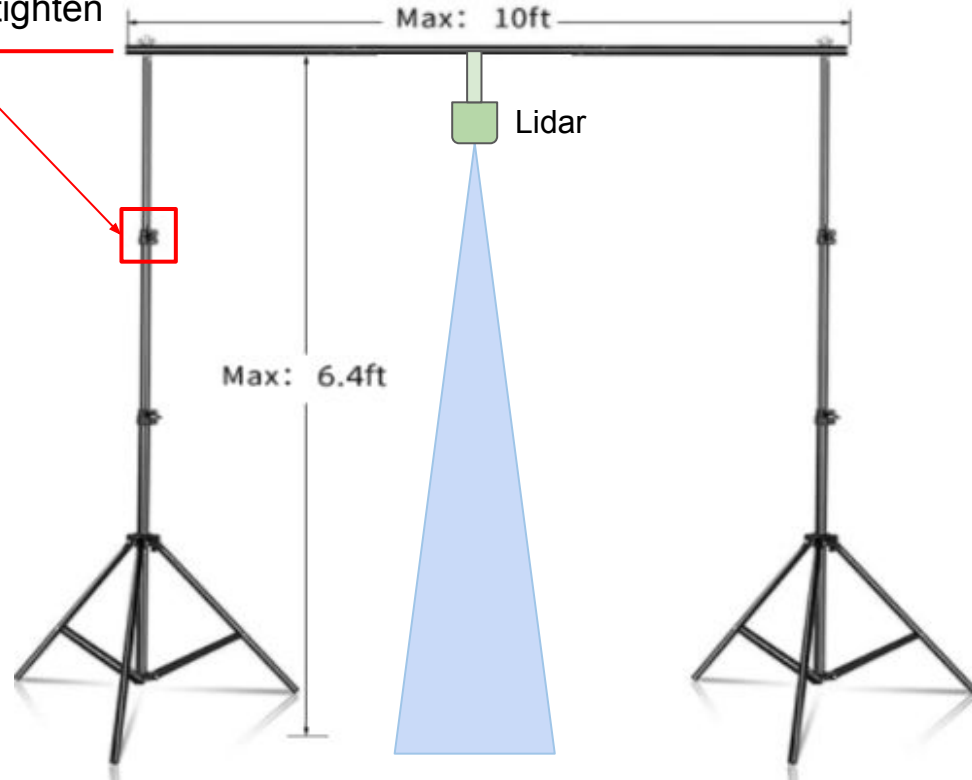
To be 3D printed by Feb 5th



- Test stand has been purchased
  - Waiting on delivery
- Lidar Mount printed out of PETG and mounts to the center of the cross beam
  - Holds lidar sensor in vertical orientation at known height

Waiting on Test stand to be delivered  
LiDAR mount printed  
Assembled by Feb 5th

Set height & tighten



\*Dimensions in mm

## Inventory

Still waiting for procurement on 1 major part

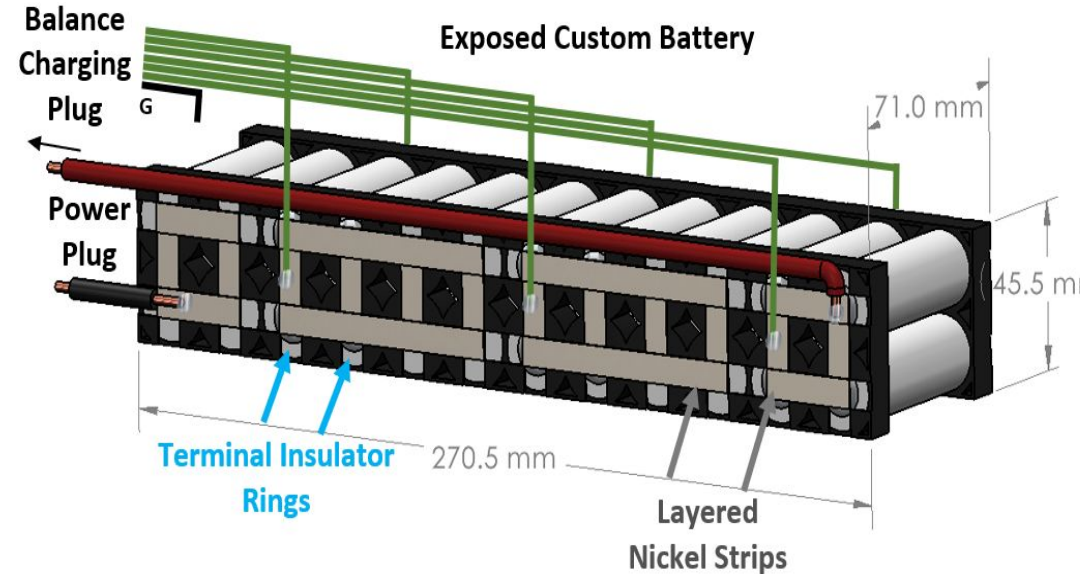
- Charging pin (delivery eta: **unknown**)

## Deliverables

1. Single simple parallel cell testing (4 cells)  
*Estimated time (1 hour)*
2. 1st iteration full battery pack (24 cells)  
*Estimated time (1 day)*

## Progress

Waiting on R2R approval to begin manufacturing



- On Hand:
  - Insulator rings
  - XT90 connectors
  - XT60 connectors
  - MT60 connectors
  - 10 AWG Wire
  - Wire Kapton tape
  - 100x LiPo battery cells
- Barriers to Progress
  - Balancing cable not delivered
  - Waiting on R2R
  - Spot welding to connect cells
  - Spot welder supplied by IRISS

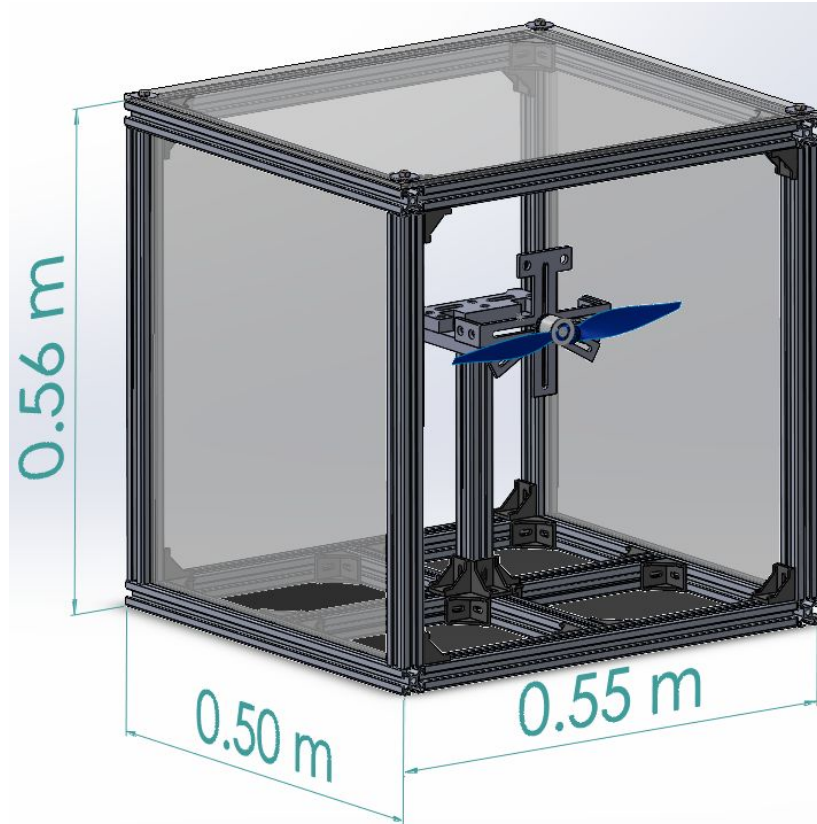


Waiting on R2R approval and balancing cable to begin manufacturing



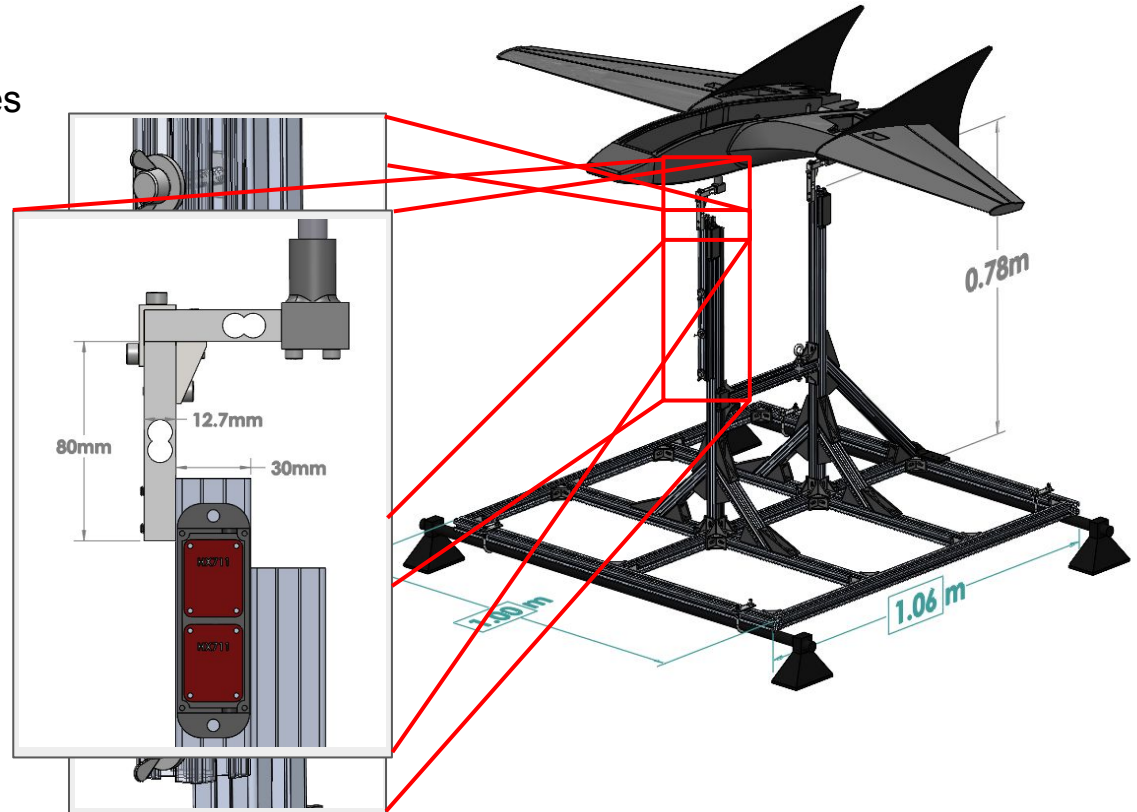
- Will be assembled by us
- Main Parts
  - Aluminum Extrusion
    - Cut to length/thread ourselves
  - 3 ¼" Acrylic Panels on sides
    - Cut to size ourselves
  - 1" wire screen for front/rear
  - Load Cell/Motor Mount Assembly
    - Lent by DBF

Cutting aluminum extrusion and assembling this week



- Main Parts
  - Aluminum Extrusion
    - Cut/assemble ourselves
  - Mounting Brackets
    - 90° aluminum
    - 45° 3D Print
    - 135° 3D Print

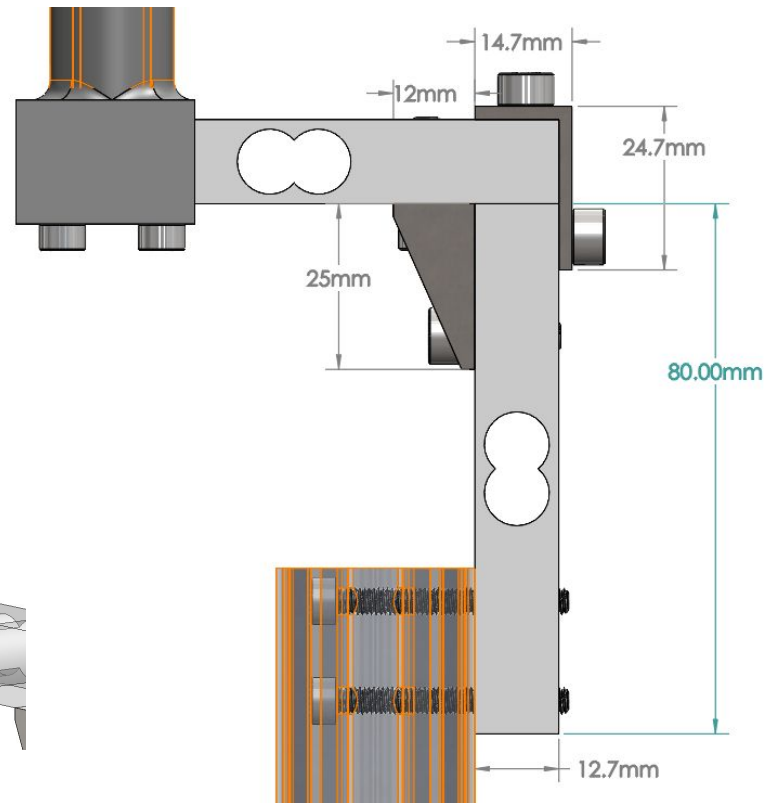
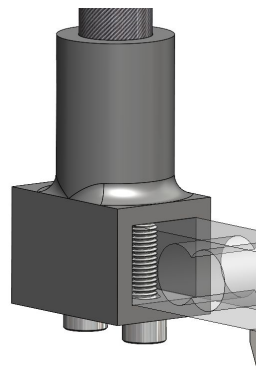
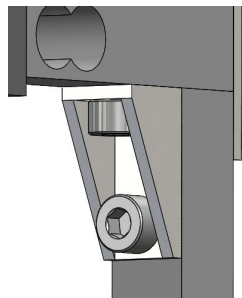
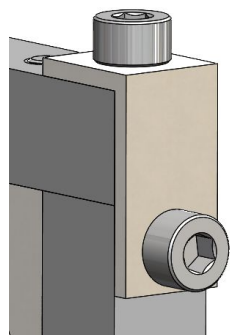
Cutting aluminum extrusion and assembling this week



## Manufacturing Sufficient Brackets

### Solution:

- Nylon Alloy 3D print bracket for cell to carbon fiber rod
- 90° inner and outer brackets milled from a low carbon steel block



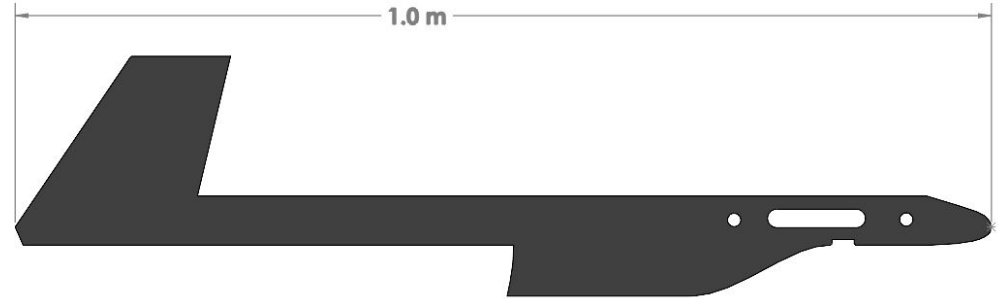




- Cut Foam Horizontal Tail
  - CNC Hot Wire Foam Cutter
  - XPS foam

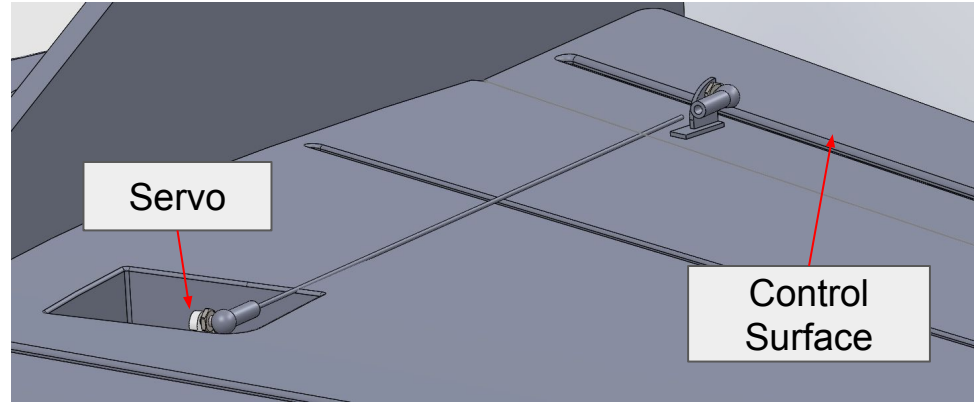


- Tail Booms
  - Laser Cut Coroplast
  - 8 mm thick



- Preliminary Dimensions (Subject to Change)
  - 1 m length
  - $S = 0.089 \text{ m}^2$
  - $V_h = 0.6$
  - $c = 0.22 \text{ m}$

- Control Surface
  - Same material and manufacturing method as horizontal tail
  - Mounted using Z-Tape
- Servo
  - Mounted inset into the horizontal tail
  - Connected to control surface same as wing
- Foam and servo still need to be purchased



Mounting example from the wing

Manufacturing of horizontal tail and control surfaces is scheduled to be completed by Feb 11th

<u>Category</u>	<u>Concern</u>	<u>Notes</u>
Dynamic Test Stand	High	Primarily safety, Milling and cutting metal, Organizing testing space, Intense data processing
Static Test Stand	Medium	Safety, Experience with DBF propulsion testing, Shop work
Custom Battery Packs	Medium	Shipping parts. Battery pack performance after manufacturing. Safety
Motor Mounts, and Loading Test	Low	Uncertainty in FEM causing failure before the required loading.
Extra Control Surface	Low	Tail booms approx. 40 in, Hot wire foam cutting, Optimization Sizing, Ardupilot parameters

**Budget**

<u>Important Parts</u>	<u>Application</u>	<u>Status</u>
LiDAR Sensor	Avionics, Testing	Received
Servos	Controls	Received
Batteries	Endurance, Testing	Received
Front and Rear Motors	Propulsion	Received
Front and Rear Propellers	Propulsion, Testing	Received
Front and Rear ESCs	Endurance/ Propulsion	Received
Ritewing Drak Kit	Structures, Testing	Received

<u>Important Parts</u>	<u>Application</u>	<u>Status</u>
Ritewing Drak - Extra Wings	Testing	Received
Nuts/Screws/Bolts etc.	Manufacturing	Received
Aluminum Extrusion	Testing, Manufacturing	Received
LiDAR Test Stand	Testing	Ordered
Load Cells	Testing	Ordered
Adhesives	Manufacturing	Ordered
Misc Battery Fabrication Materials*	Testing	Received

\*Battery brackets for fabrication are about a month or more delayed



## VORTEX Budget Breakdown:

Total Budget: \$5,000.00

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Confirmed Purchases: \$3,560.00

Pilot Lab Deposit: -\$200.00

(assuming is returned)

Estimated Future Costs: \$400.00

Contingency Budget: \$1000.00 \*

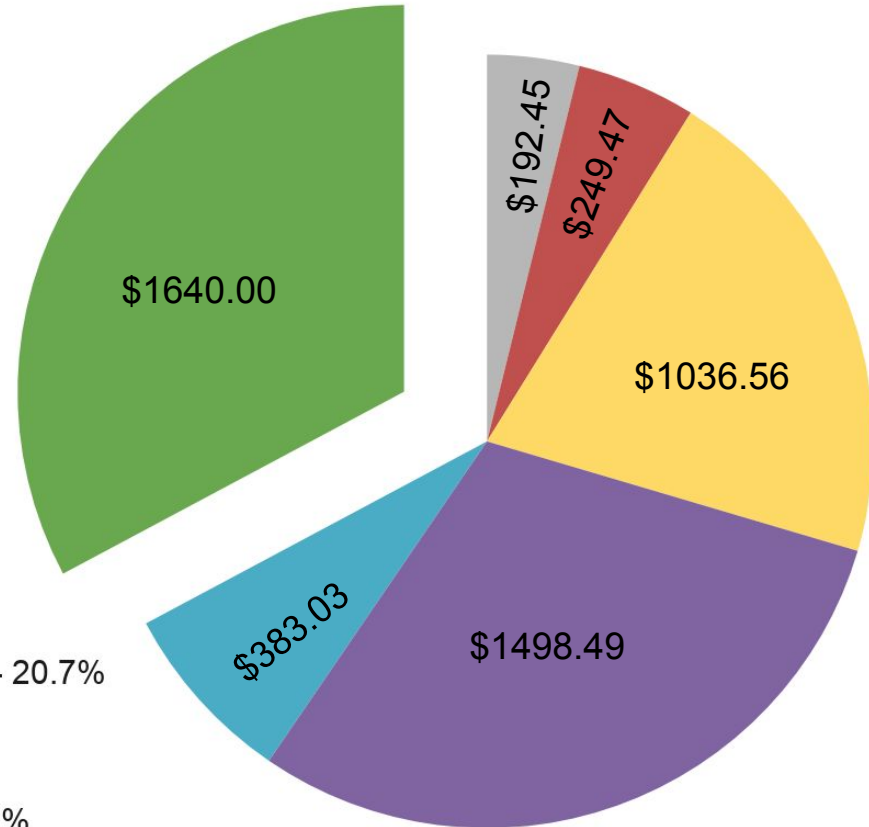
Total Expenses: \$4,760.00

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**Estimated Final Balance: \$240.00**

\***Note:** Estimated final balance assumes entire contingency/complication budget is used.

- Avionics - 3.8%
- Controls - 4.3%
- Endurance/ Propulsion - 20.7%
- Structures - 30.0%
- Testing - 7.7%
- Remaining Funds - 29.5%



**Thank You!**

**Questions?**

# Backup Slides



## Options Considered

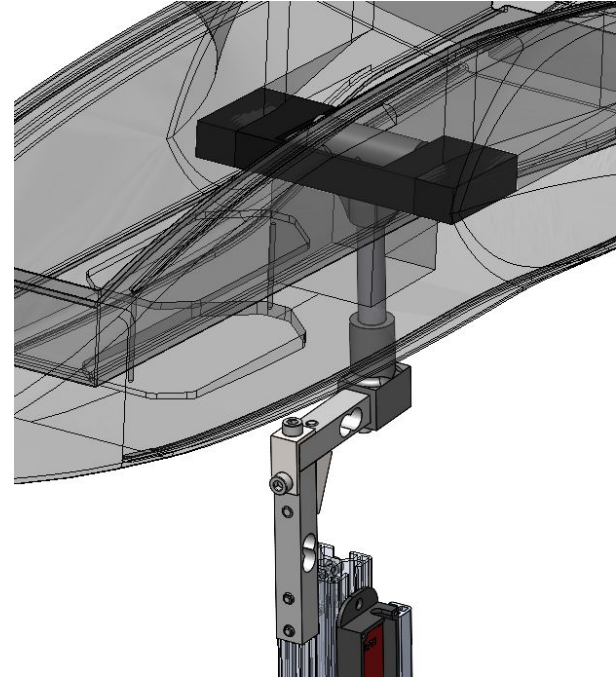
- Static Canard
- Static Tail
- Tail with Elevator

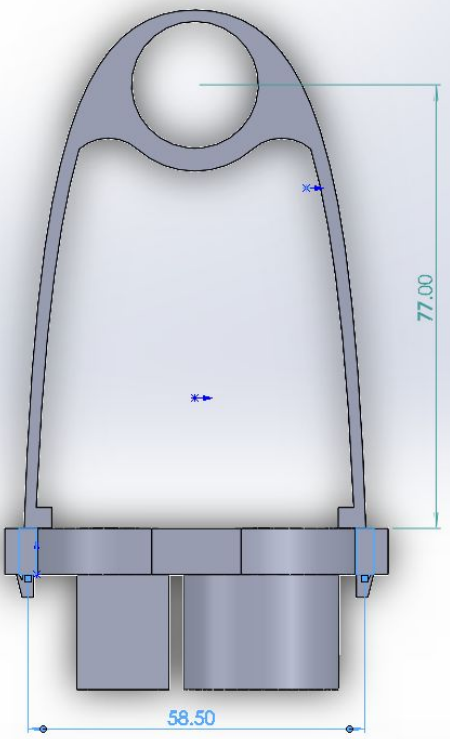
## Tail with Elevator won out

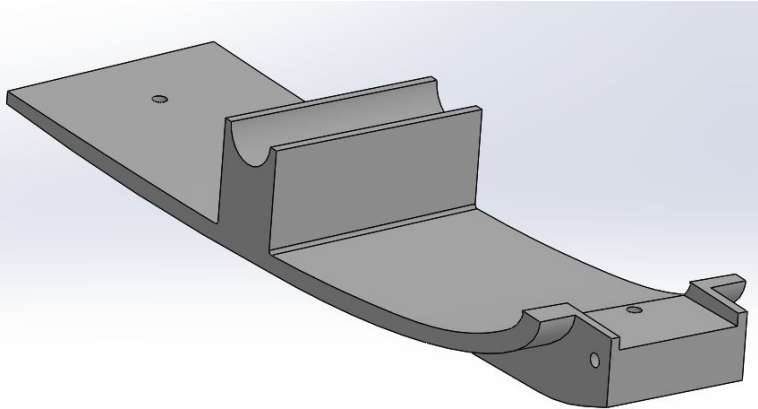
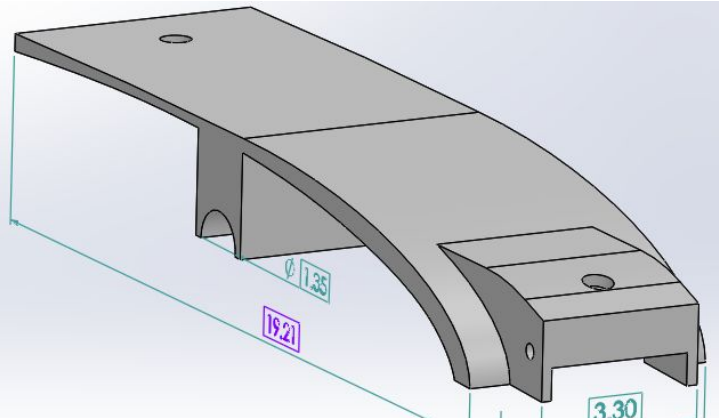
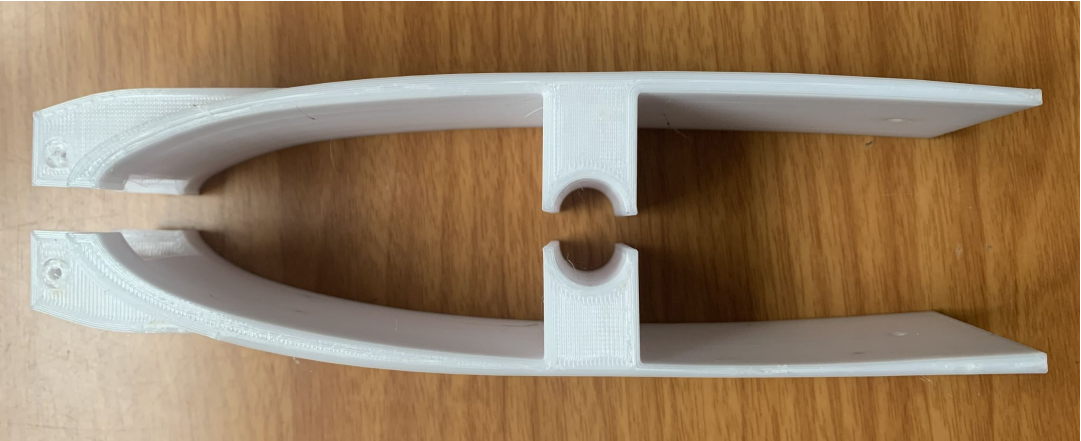
- Reduces elevon deflection
- Used by IRISS
- Heavier than Canard though
- Easy to manufacture

Criteria	Things to Consider	Criteria Weight
Lift and Drag Performance	<ul style="list-style-type: none"> <li>• Trim Lift Deficit Calculations</li> <li>• Estimate Drag, NACA airfoil, Parasite, Induced, etc.</li> <li>• Effect on Elevon deflection at trim</li> </ul>	25%
Stability	<ul style="list-style-type: none"> <li>• Static Margin Calculations</li> <li>• Trim Moment Deficit Calculations</li> <li>• Stall performance</li> <li>• Static, dynamic stability</li> </ul>	30%
Weight	<ul style="list-style-type: none"> <li>• Weight of supplementary components                             <ul style="list-style-type: none"> <li>◦ Servos, Spars, Foam</li> </ul> </li> <li>• Shift in center of gravity</li> </ul>	15%
Complexity	<ul style="list-style-type: none"> <li>• Supplementary Components</li> <li>• Electronics</li> <li>• Structure required</li> <li>• Manufacturability                             <ul style="list-style-type: none"> <li>◦ Materials and Methods</li> </ul> </li> <li>• Design Optimization                             <ul style="list-style-type: none"> <li>◦ Expected effort to optimize the design</li> </ul> </li> </ul>	30%

- Internal bracket allows spar to slide forwards and backwards to change angle
- Horizontal carbon fiber rod slides inside internal bracket
- Internal bracket will be 3D printed out of PETG











The purpose of the early manufacturing stages with regards to the VORTEX project is to both build and test individual subsystems.

1. Assemble basic functioning subsystems
2. Assemble testing apparatuses
3. Test and simulate realistic performance against modeled performance

Moving forward, each subsystem will be iteratively improved to meet desired performance. Full system testing can begin.

Subsystem	Testing Equipment
Autonomy	<ul style="list-style-type: none"><li>● Ardupilot</li><li>● Pixhawk</li><li>● LIDAR Test Stand</li></ul>
Structures	<ul style="list-style-type: none"><li>● Dynamic Test Stand</li></ul>
Propulsion	<ul style="list-style-type: none"><li>● Static Test Stand</li><li>● Construction Battery</li><li>● Dynamic Test Stand</li></ul>

## Key

Software (in-house)

Custom Hardware (in-house)

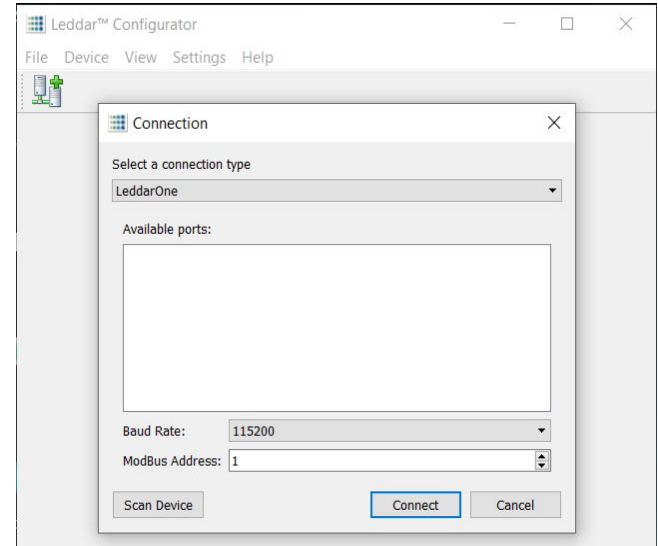
Borrowed Hardware

## Testing the purchased LiDAR sensor to verify 10cm accuracy

### LeddarTech Configurator

- Exports data to .txt file
- USB to UART cable to laptop

### LeddarTech Configurator



### LeddarOne Sensor

