



Smead Aerospace
UNIVERSITY OF COLORADO BOULDER



Deep-Space Orbital Telecommunications

Sam Taylor, Tristan Liu, Caelan Maitland, Forrest Jordan, Hunter Rohlman, N. Sebastian Damm, Doug Brough, Conner Lewis, Buck Guthrie, Jennifer Gurtler



Mission Statement

Project DOTCOM is a research-heavy **system modeling** assignment. In this, we explore the functionality and viability of a communications network architecture between the Earth and Moon. The purpose of this project is to **develop software models** to design and optimize a **Lunar communications network**, packaged through **Model-Based System Engineering**.



Table of Contents

1 Project Purpose and Objectives

1.1 CONOPS

1.2 Critical Project Elements

1.3 Performance Targets

1.4 Baseline Design

1.5 FBD

2 Project Schedule

3 Manufacturing

3.1 STK Architecture Model

3.2 Network Link Budget

3.3 Network Capacity Model

3.4 ION-DTN Demonstration

3.5 SysML Systems Engineering Model

4 Budget



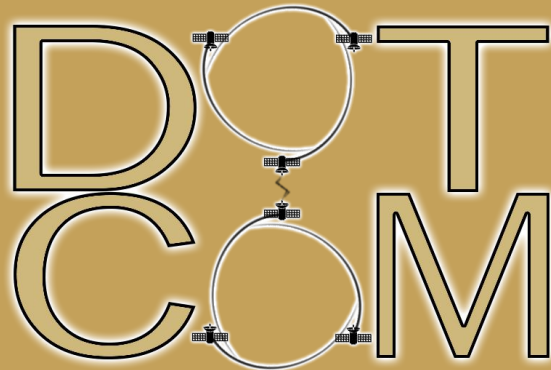
Project Purpose & Objectives

Project Purpose & Objectives

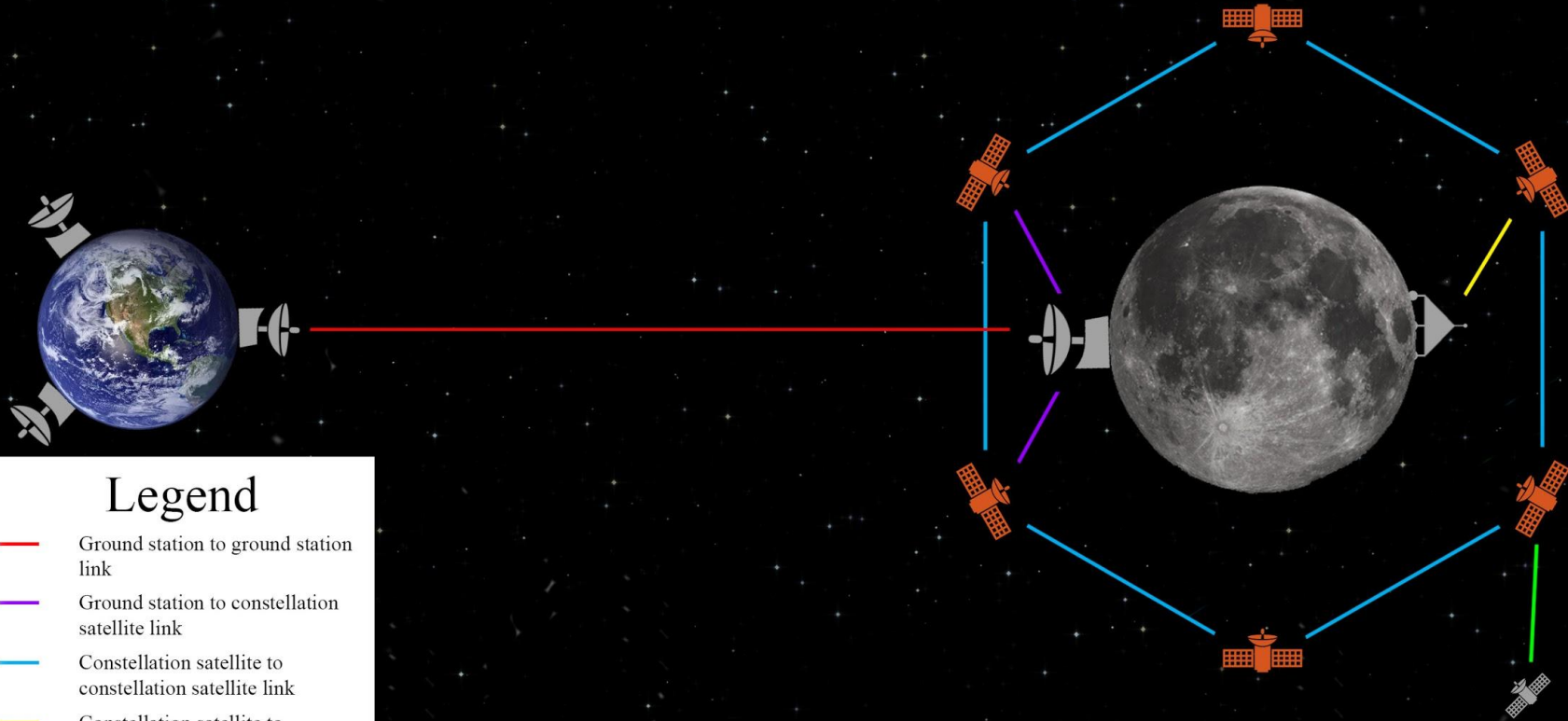
Project Schedule

Manufacturing

Budget



System CONOPS



Legend

- Ground station to ground station link
- Ground station to constellation satellite link
- Constellation satellite to constellation satellite link
- Constellation satellite to ground vehicle link
- Constellation satellite to orbital vehicle link



Critical Project Elements

Designation	CPE	Critical Characteristics
CPE-1	Network Protocol	Structured data transmission methodology that allows for high speed reliable communications from node to node.
CPE-2	System Link Budget	The project will meet certain data-relay rates for communication between all communication nodes.
CPE-3	Relay Stations	Allows for direct access to communications between Earth and The Moon.
CPE-4	Satellite Constellations	Construction of ideal constellation architecture around each planetary body to satisfy coverage requirements.



Performance Targets



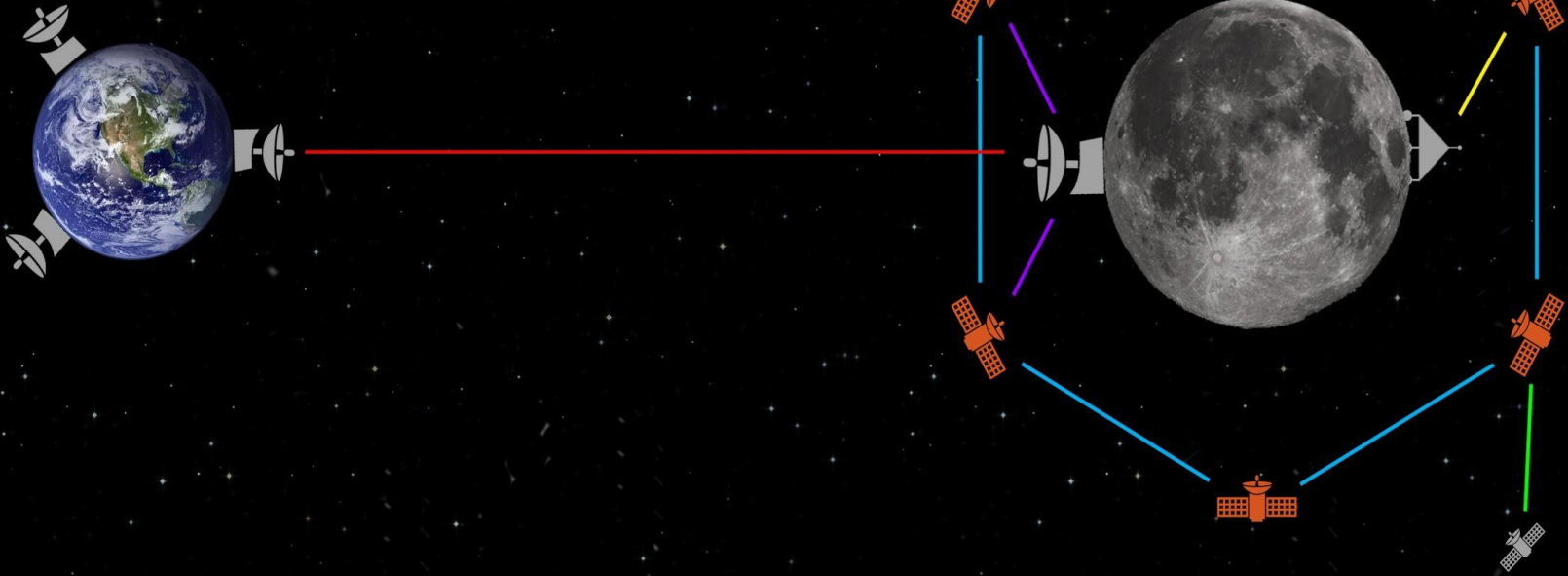
Functional Requirement	Design Requirement
FR 1 Area Coverage	99% telecommunications coverage in Lunar orbit
	99% telecommunications coverage on Lunar Surface
FR 2 Endpoint Support	5+ nodes for simultaneous 'real time' communications on Lunar surface
	10+ nodes for non simultaneous (within 6 hrs) communication on Lunar surface
	10+ nodes for simultaneous 'real time' communication in Lunar orbit
	20+ nodes for non-simultaneous (within 6 hrs) communication in Lunar orbit
	Earth-Moon data rates: 500 Mbps threshold, 5 Gbps objective
FR 3 Compatibility	Network is compatible with and ensures the safety of existing and future infrastructure



Baseline Network Design

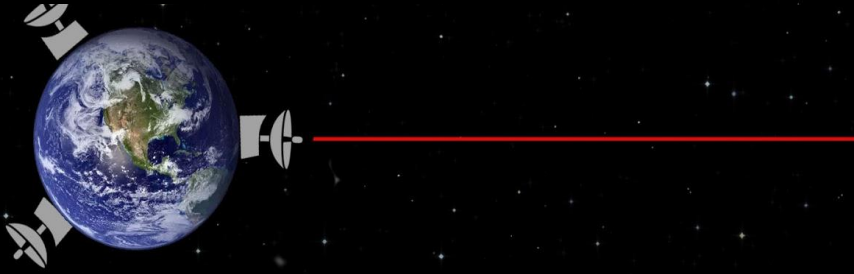
Deep Space Relay Stations

- 3 Earth Ground Stations 120° apart
- 1 Lunar Ground Station at center of “light” side



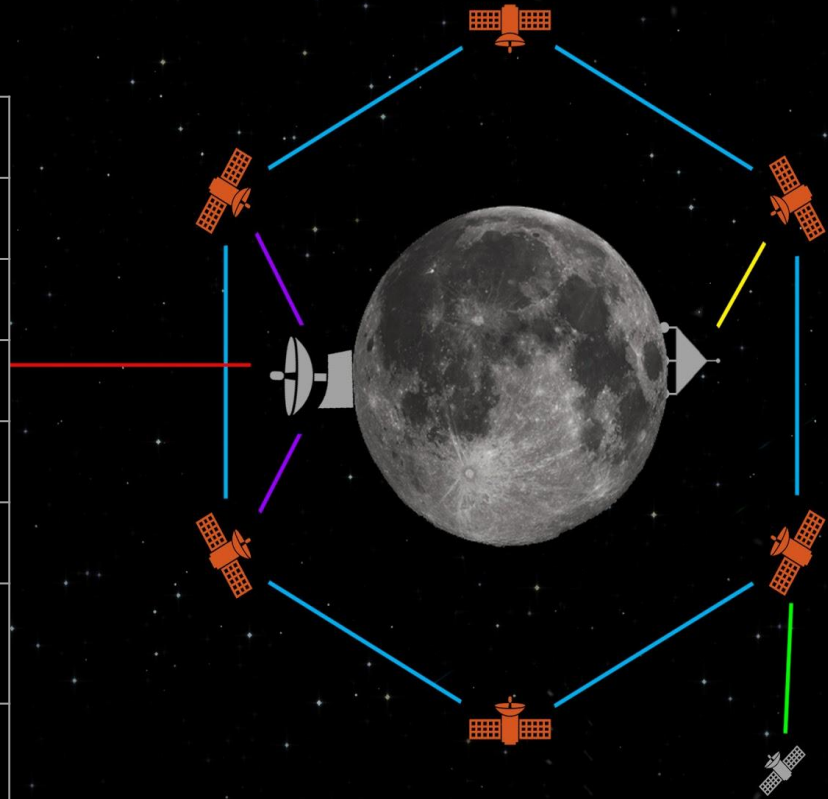
Satellite Constellation Architecture

- Walker-Delta configuration
 - ~53 deg inclination
 - 6/6/4 configuration
 - MEO equivalent altitude



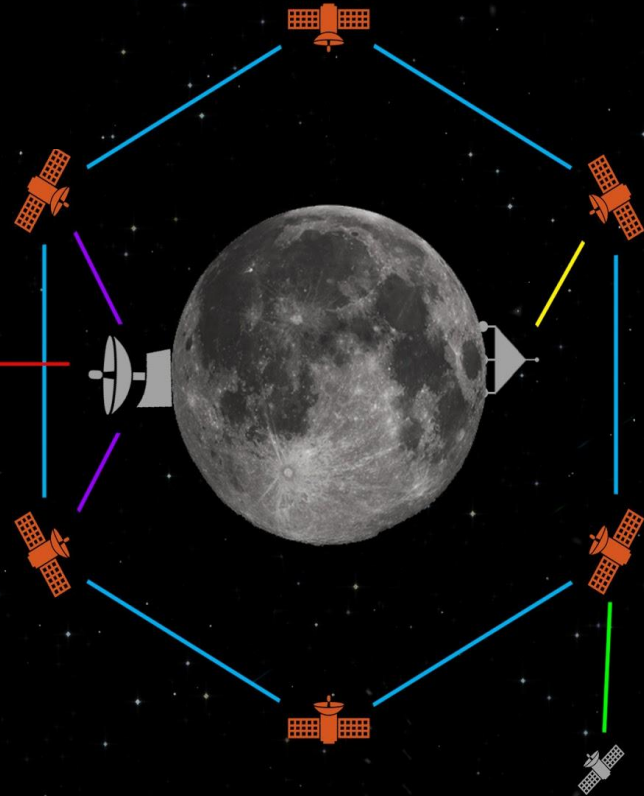
Inter-Constellation Satellite Link Budget

Link Parameters	Value
Max Range (km)	11,024 km
Frequency (GHz)	26
Antenna Size (D)	1 m
Transmit Power (W)	30 W
Data Rate (Mbps)	500
Receive System Noise Temperature (K)	700 K
Required Eb/No (dB) [BPSK Modulation, BER = 10^{-7}]	11 dB
Required Design Margin	3 dB

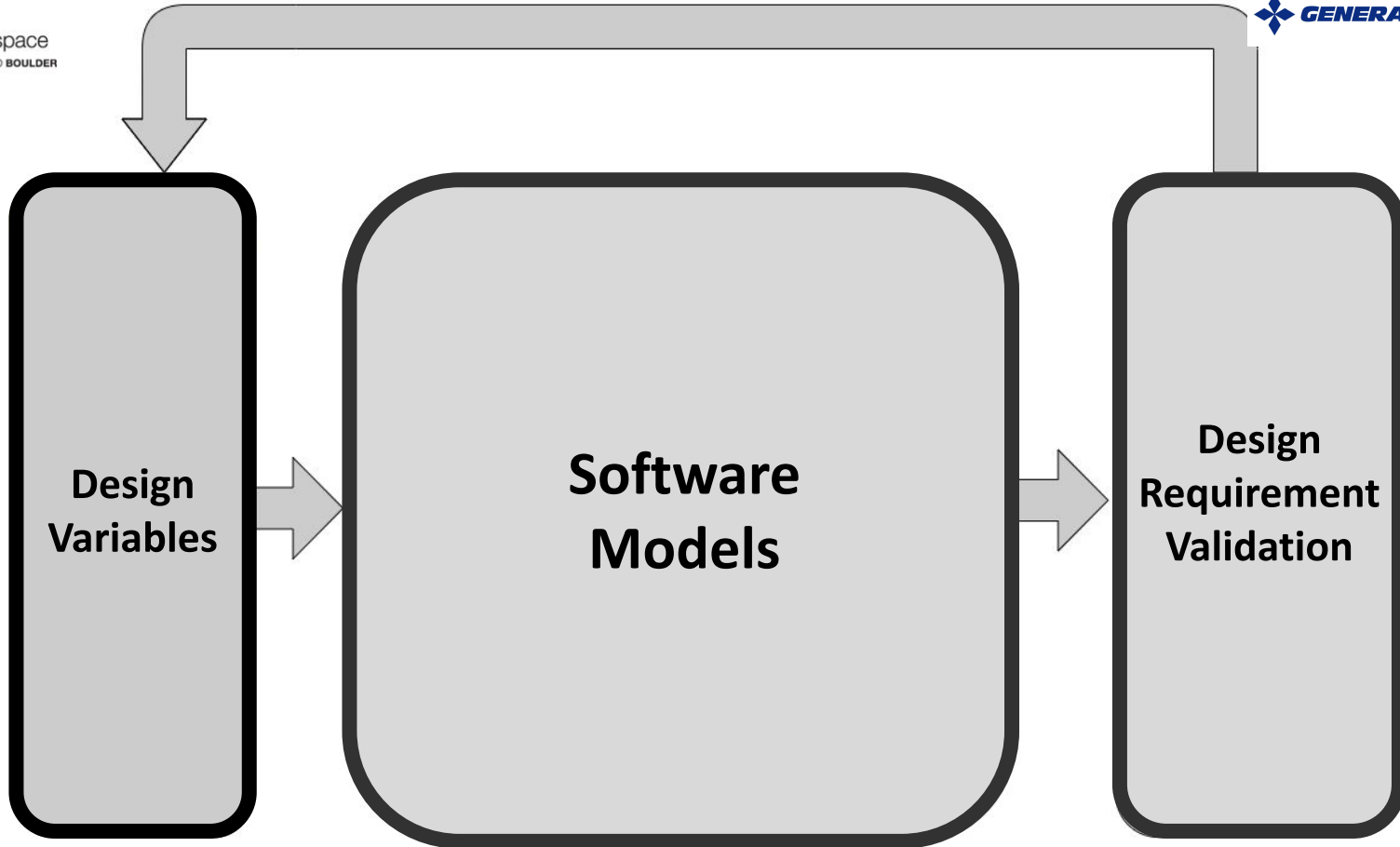


Network Protocol

- Interplanetary Overlay Network (NASA JPL)
 - Implements Delay Tolerant Networking (DTN)



User application, e.g., data manager			
CFDP (unacknowledged mode)		AMS messaging	
UT adapter		Remote AMS bridging	
BP DTN routing			
Convergence layer adapters			
LTP		TCP, BRS, UDP, DGR	
encapsulation packets		IP Internet routing	
AOS	Prox-1	802.11	Ethernet
R/F, optical		wire	



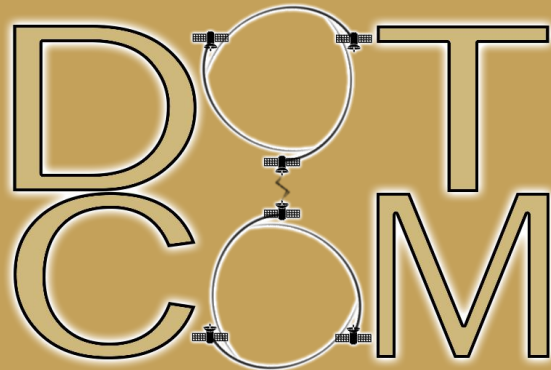
Project Schedule

Project Purpose & Objectives

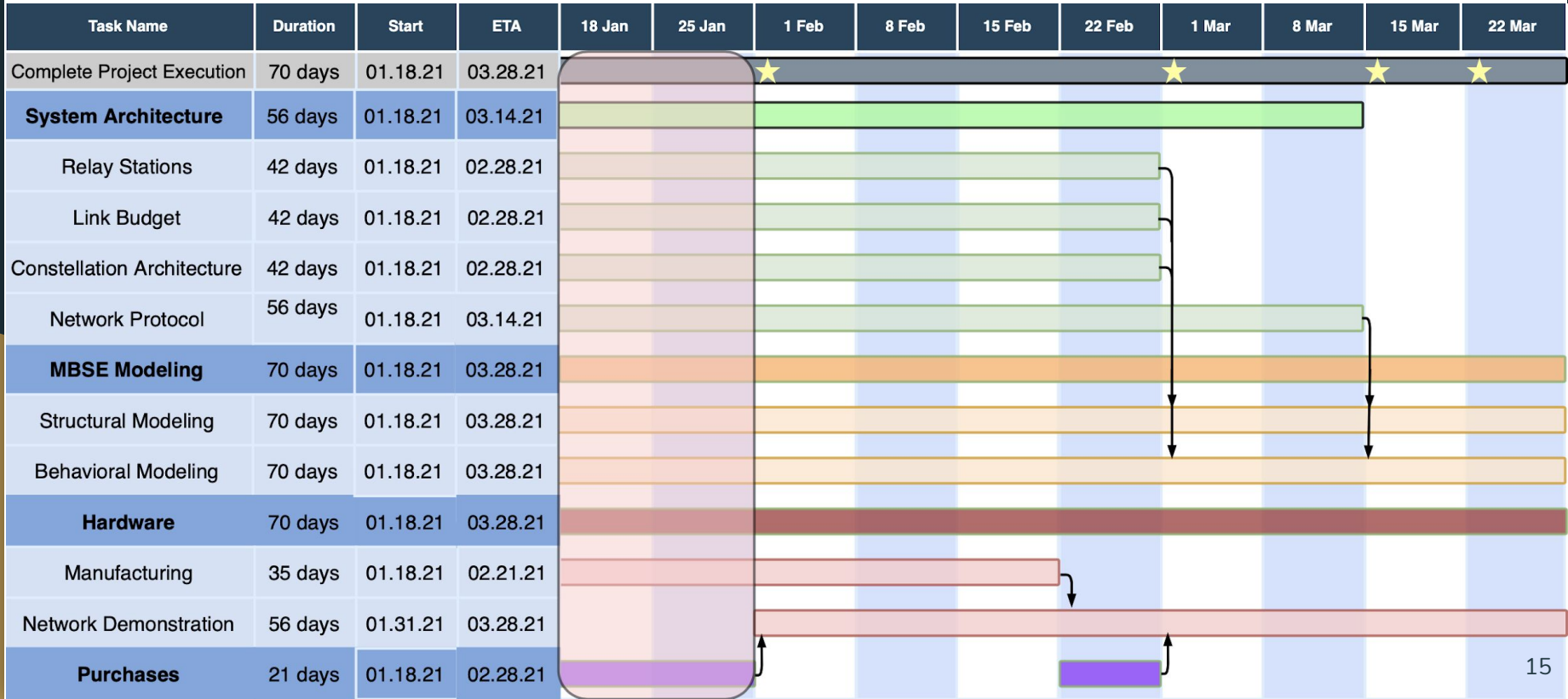
Project Schedule

Manufacturing

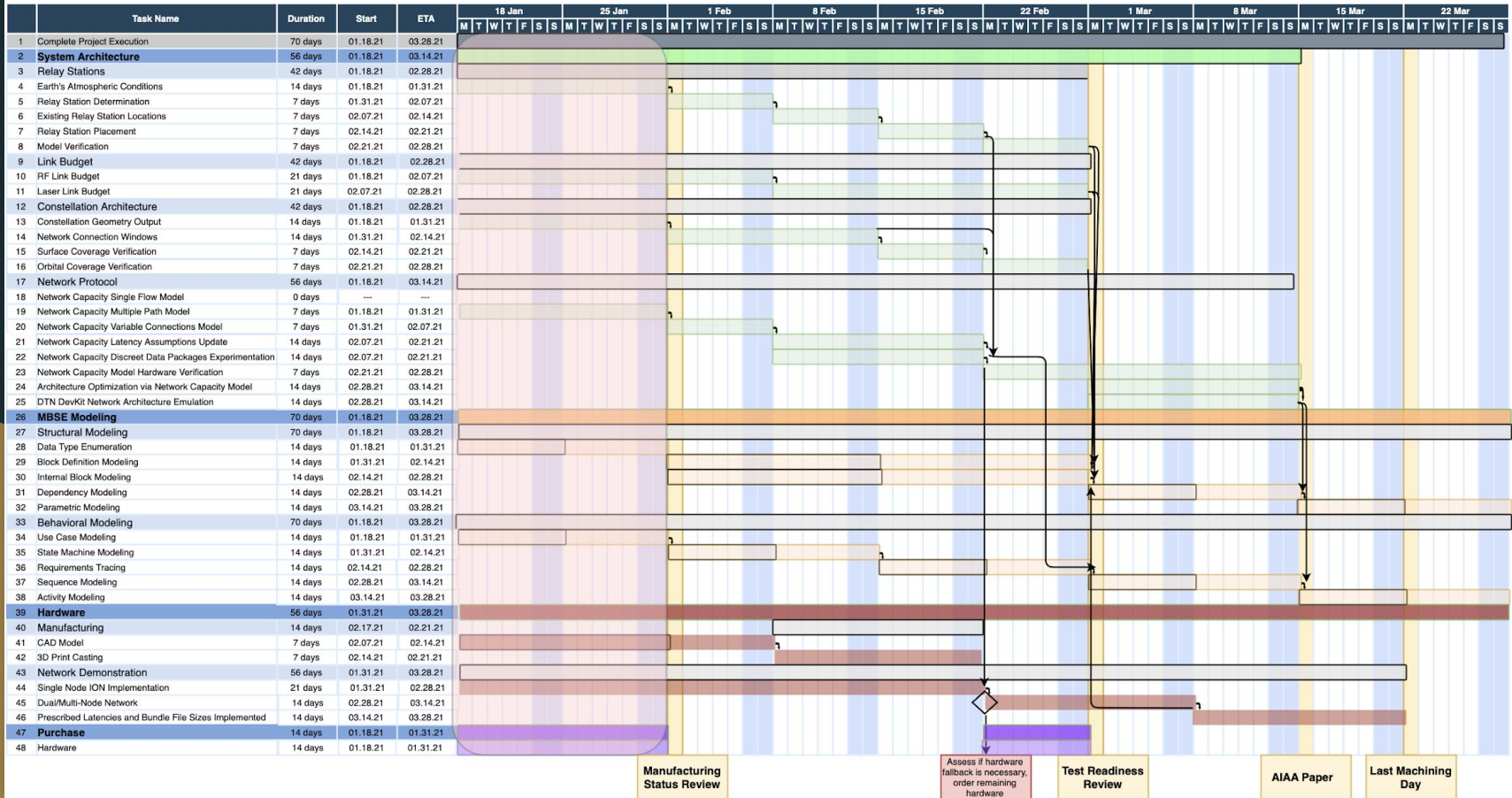
Budget



Schedule Overview



Detailed Schedule



Week 14:
Senior Design Symposium

Week 15:
Spring Final Review
College of Engineering Expo

Week 16:
Spring Final Review
Project Final Report
Final Check Our Form

Manufacturing Status Review

Assess if hardware fallback is necessary, order remaining hardware

Test Readiness Review

AIAA Paper

Last Machining Day

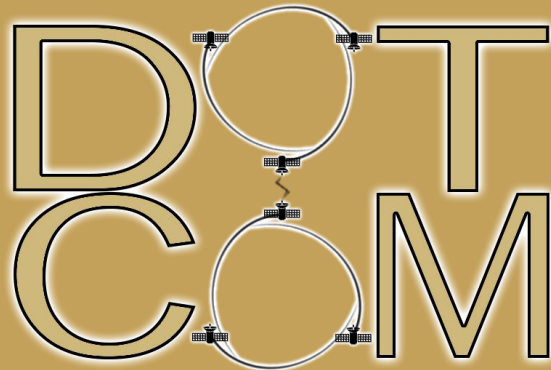
Manufacturing

Project Purpose & Objectives

Project Schedule

Manufacturing

Budget





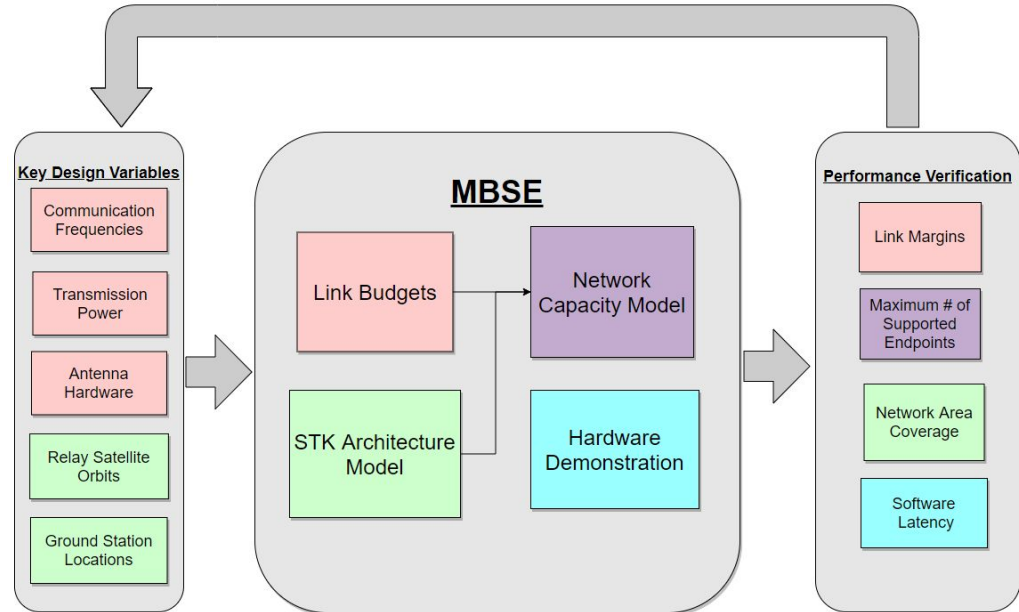
Manufacturing Tasks

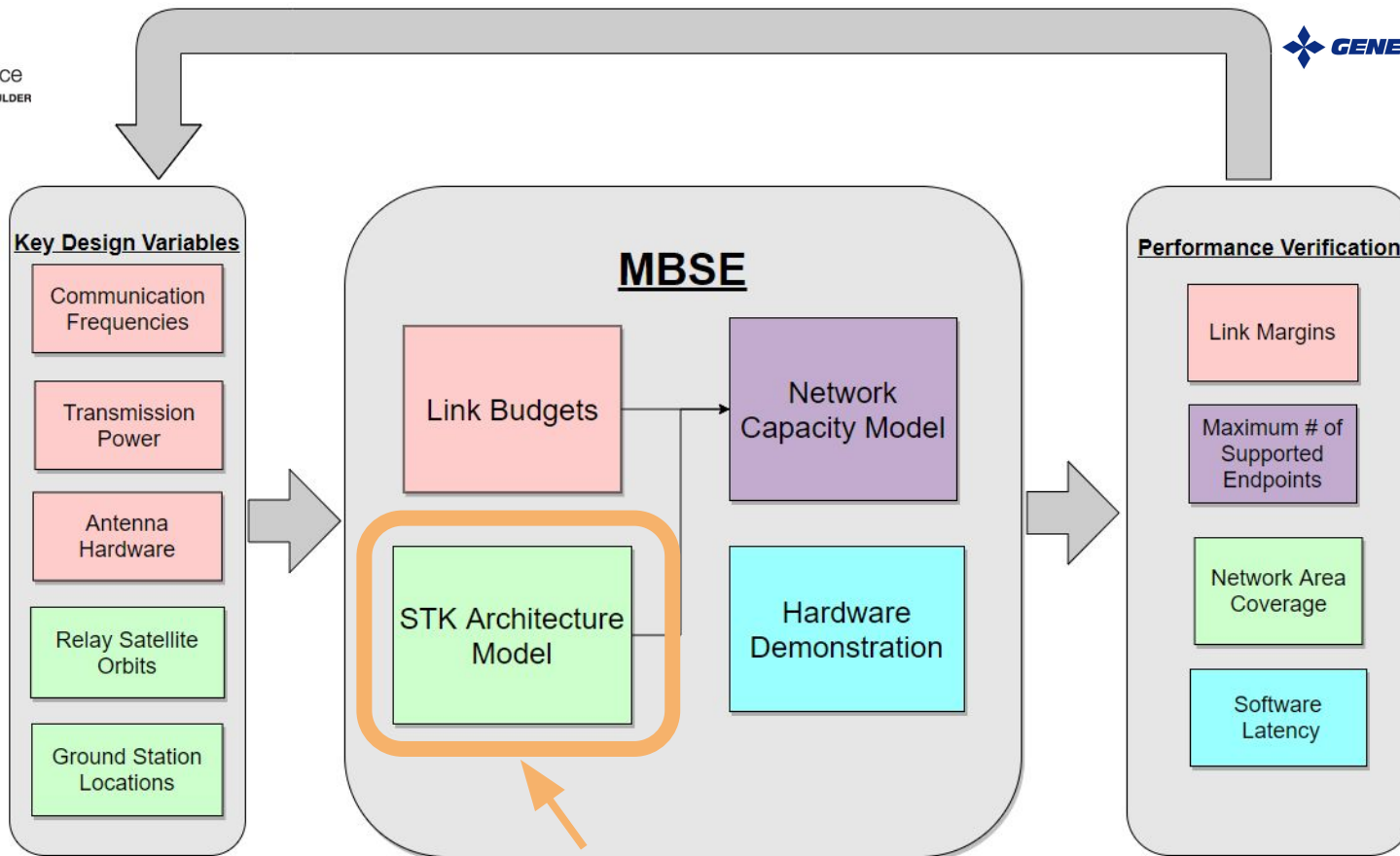
To be **manufactured**:

- 1) STK Architecture Model
- 2) Network Link Budgets
- 3) Network Capacity Model
- 4) MBSE System Model

To be **purchased**:

- 5) Hardware Demonstration







STK Architecture Model



Purpose: Determine the ground station configuration and satellite constellation architecture necessary to enable 99% Lunar coverage and “real time” communication between the Earth and the moon.

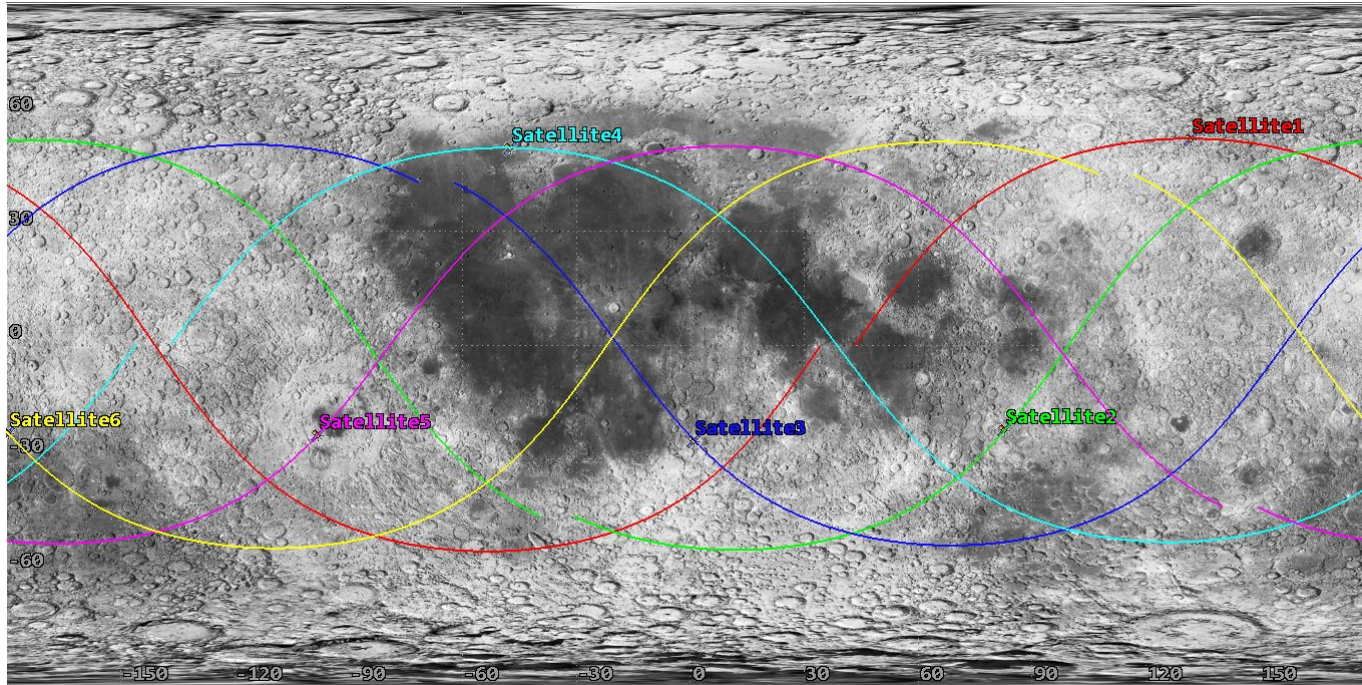
Model Logistics:

- Connection window simulation using Orbital STK
- **Inputs:** ground station locations (based on findings from atmospheric attenuation), satellite orbit geometry, satellite downlink beamwidth
- **Outputs:** Connection window illustration, 99% lunar coverage verification

Model Validation: Existing architecture verification (Deep Space Network), sensitivity analysis, and available literature/research.



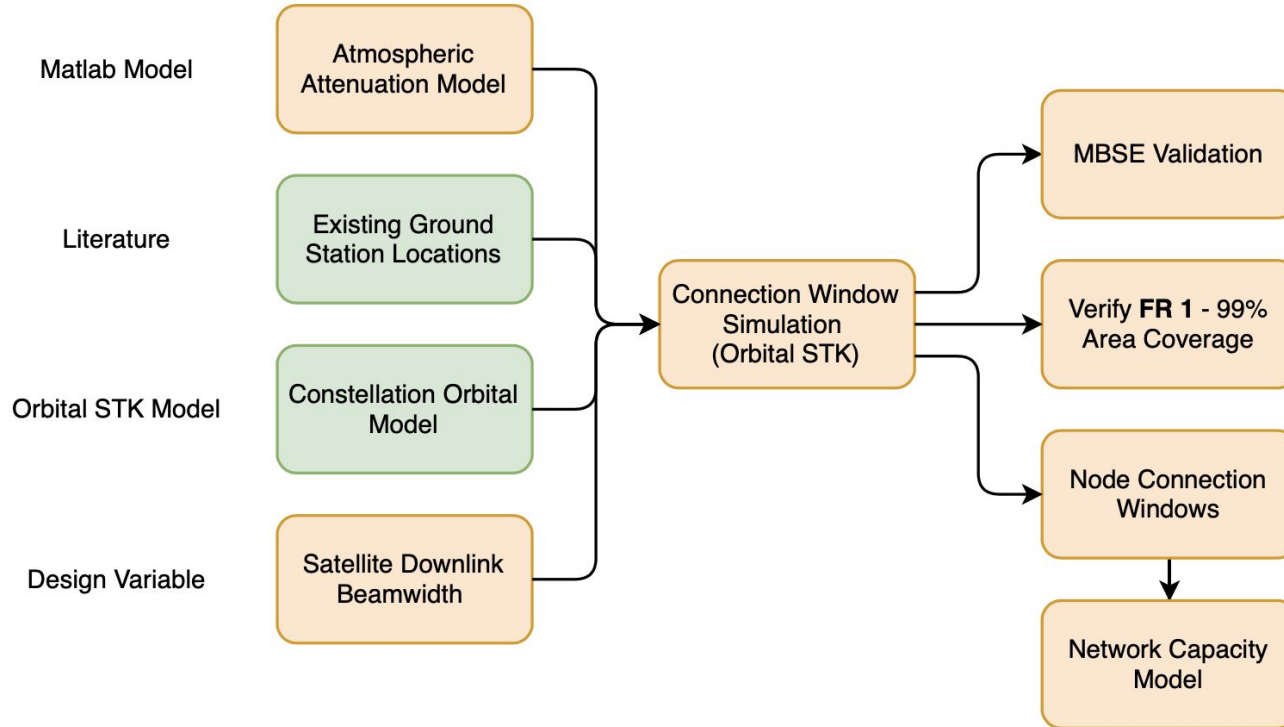
STK Constellation Orbital Model

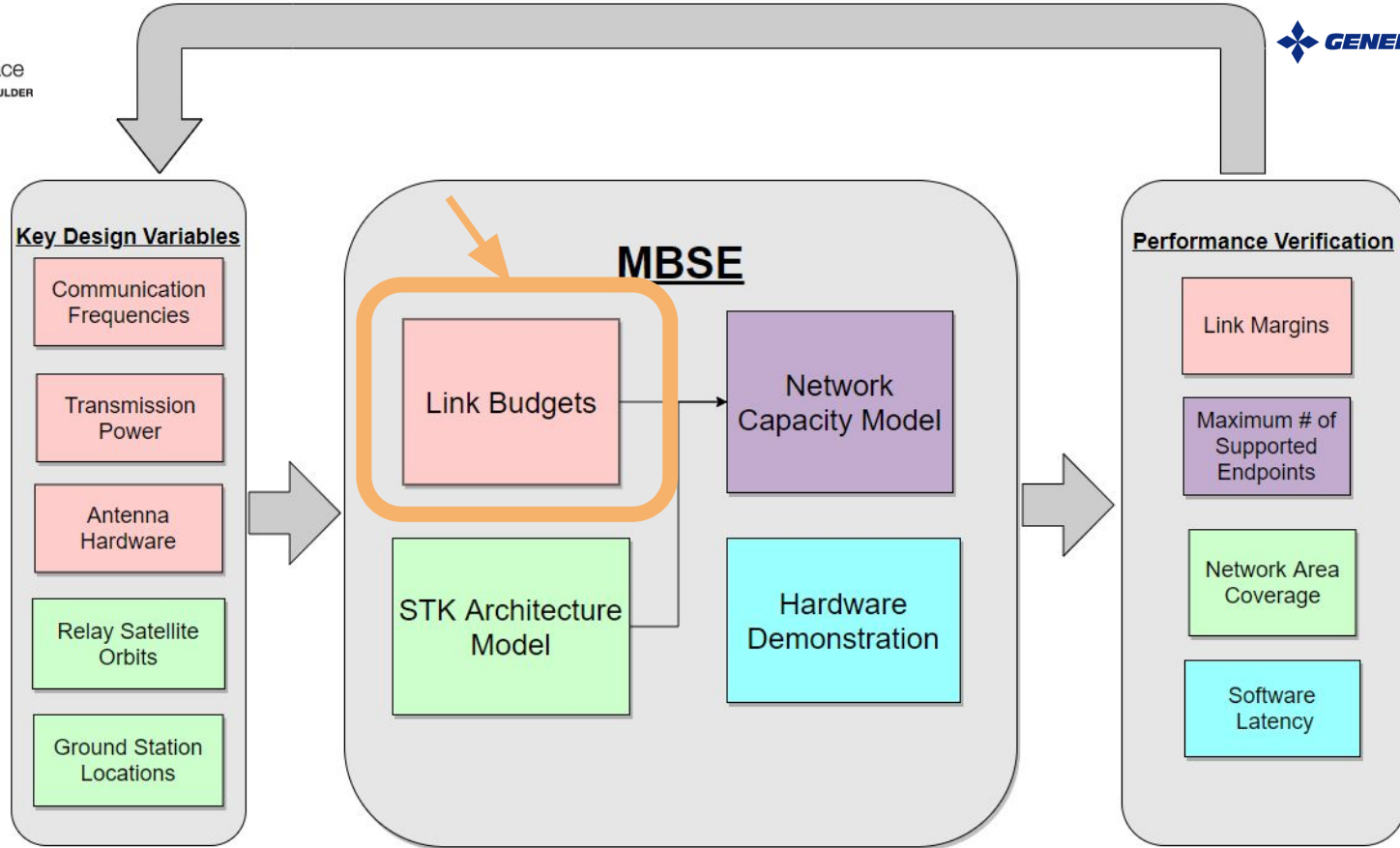


*Pictured above (baseline design) - Walker J.G., Continuous Whole-Earth Coverage by Circular-Orbit Satellite Patterns, Royal Aircraft Establishment, September 23, 1977



STK Architecture Build Progress







RF Link Model

Purpose: Process necessary inputs to create a successful link (positive link margin)

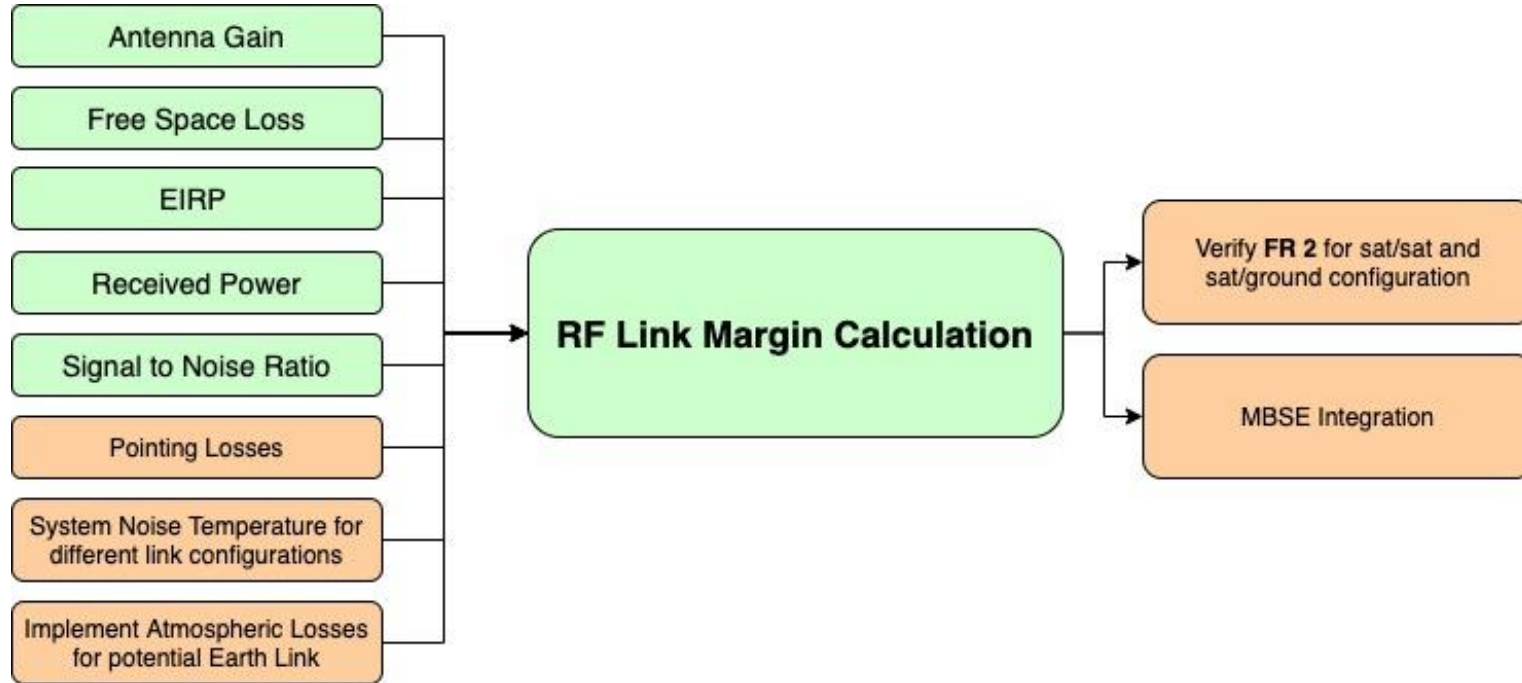
Model Logistics:

- Programed using MatLab
- **Inputs:** Range, Frequency, Antenna Size, Transmit Power, Data Rate, System Noise temperature, Required Energy per bit to noise ratio (E_b/N_0) for desired BER, Required Design Margin
- **Outputs:** Link Margin

Model Validation: Available literature and research, sensitivity analysis

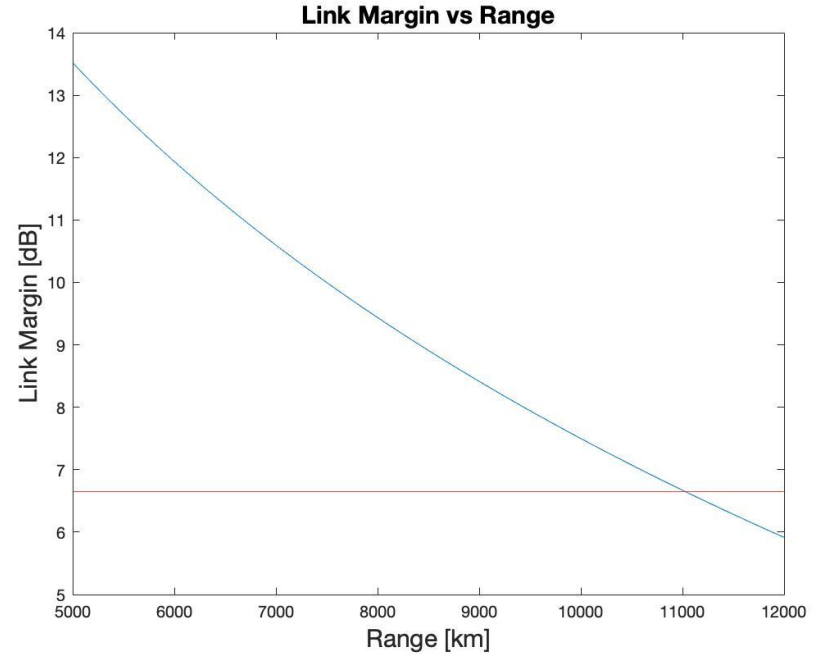
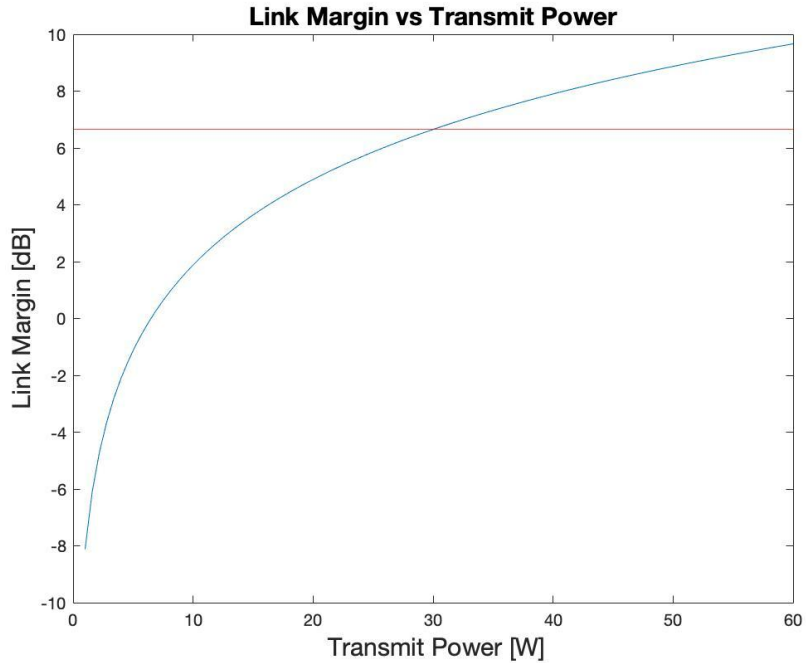


RF Link Model Progress





Baseline Sensitivity Analysis Intersatellite Link





Laser Link Model

Purpose: Process necessary inputs to create a successful link (positive link margin)

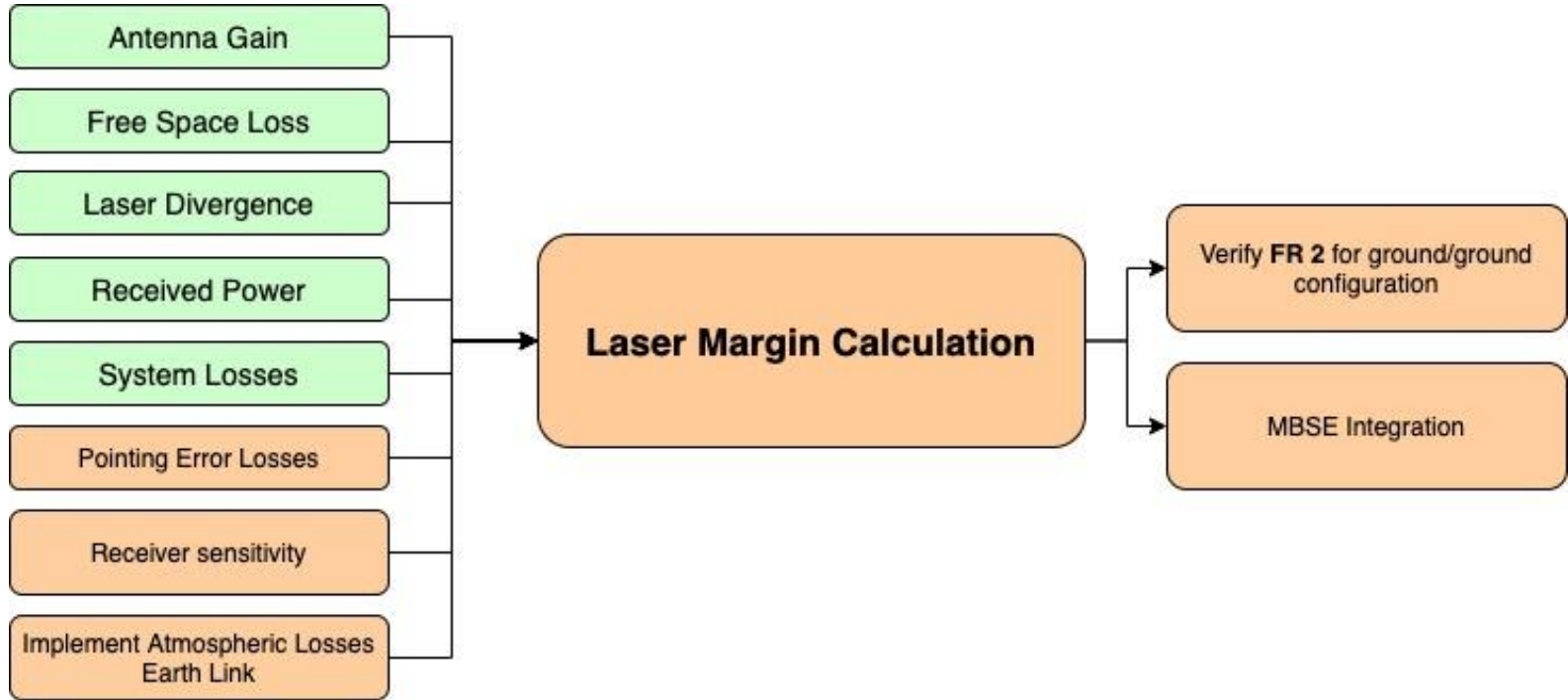
Model Logistics:

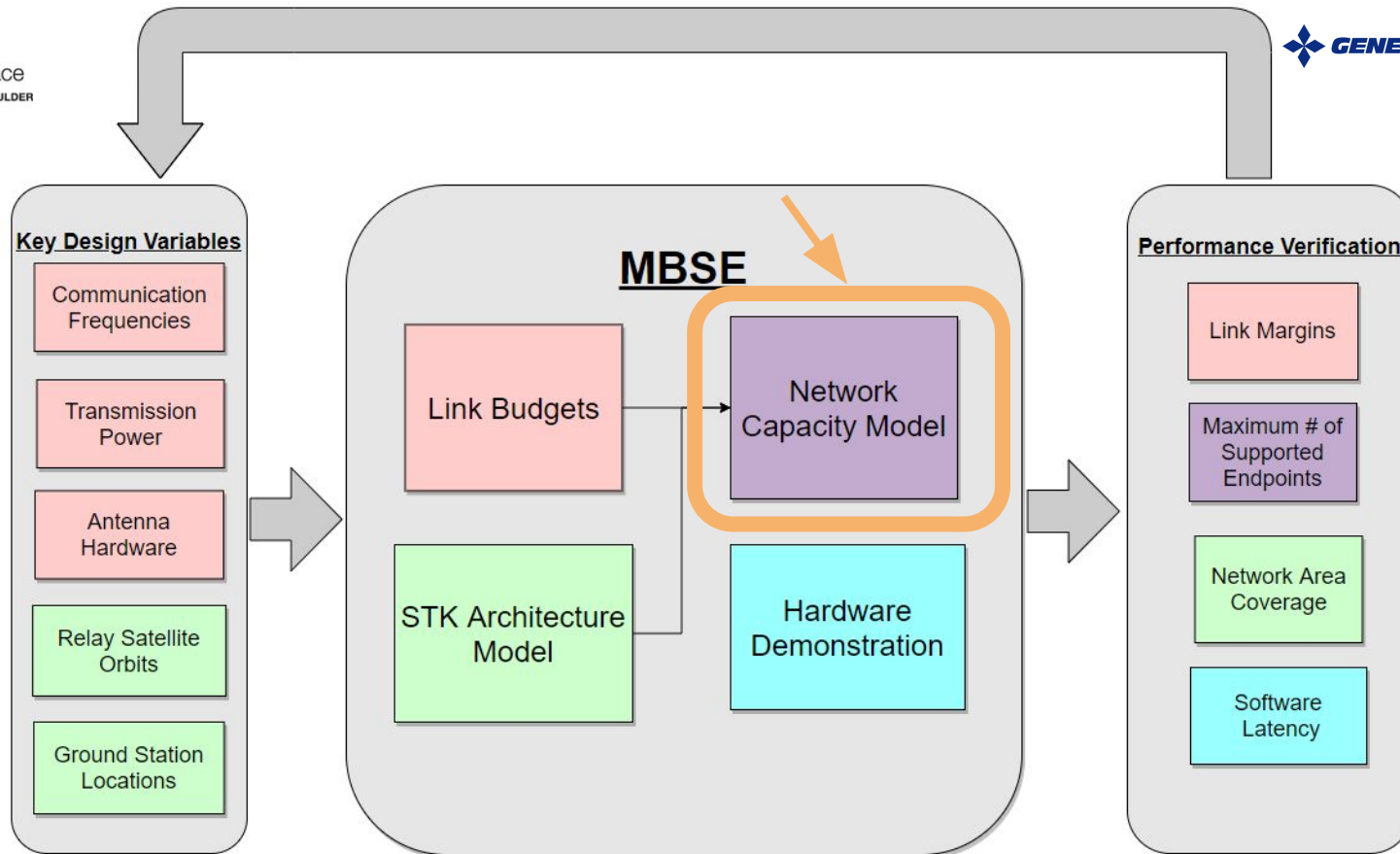
- Programed using MatLab
- **Inputs:** Range, Frequency, Antenna Size, Transmit Power, Data Rate, System Losses (Atmospheric, transmit, receiver)
- **Outputs:** Link Margin

Model Validation: Available literature and research, sensitivity analysis



Laser Link Model Progress







Network Capacity Model

Purpose: Verify that the network can support the required number of endpoints

Model Logistics:

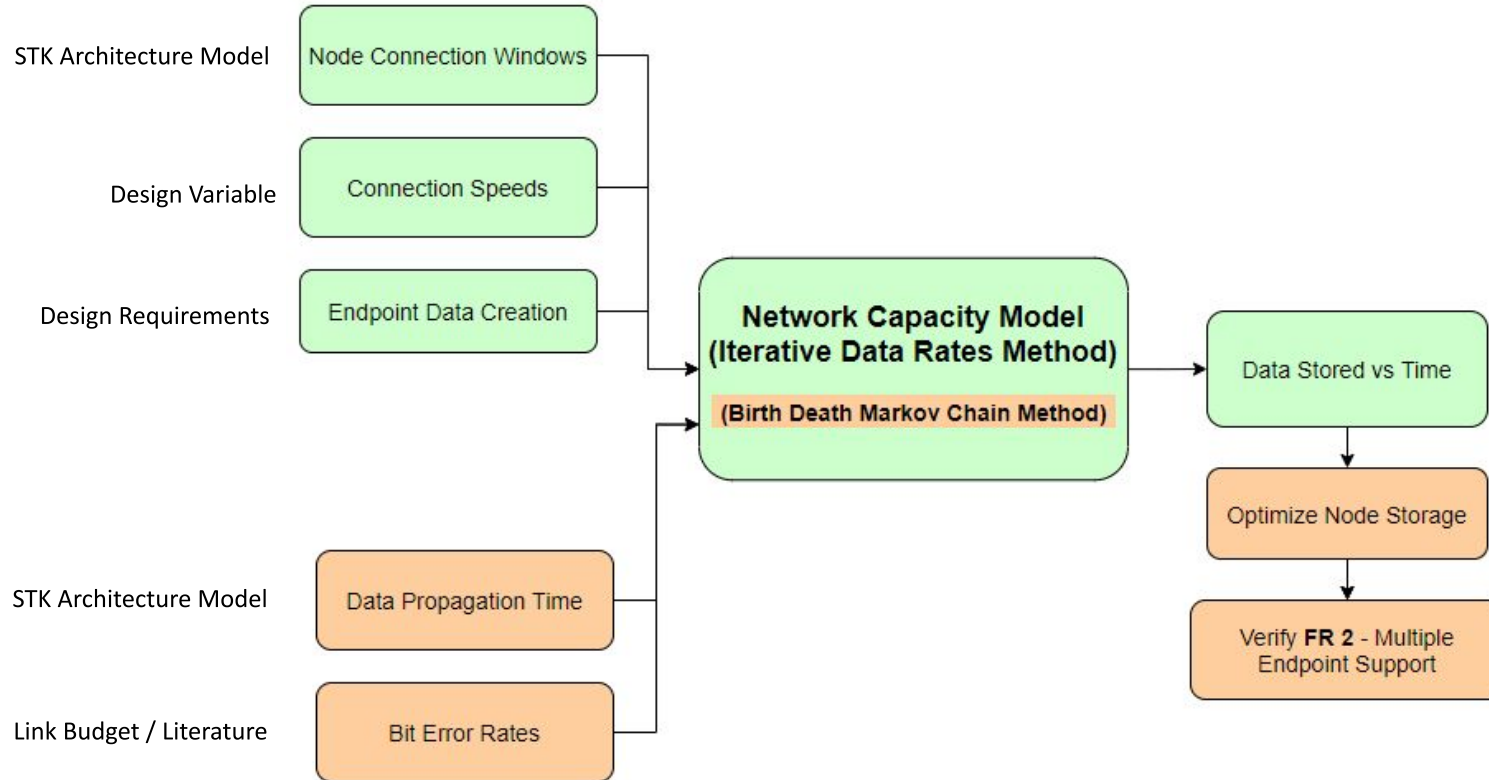
- Iterative data rates computation method performed using MatLab
- **Inputs:** node connections, link data rates, endpoint data requirements, bit error rates, data propagation delay
- **Outputs:** Data stored in each node over time

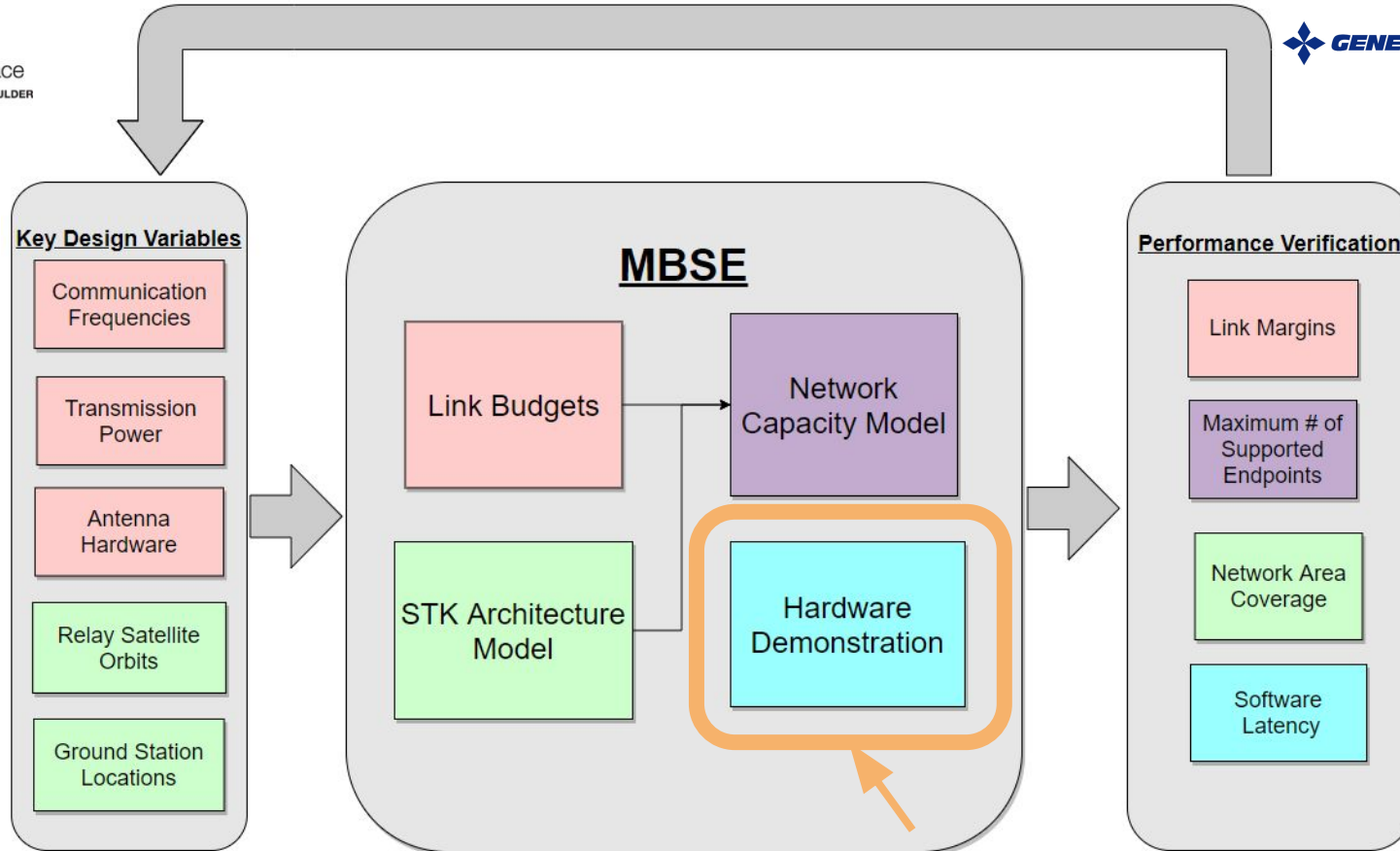
Model Validation: Benchmarking and hardware verification test





Capacity Model: Build Progress







Hardware Demonstration



Purpose: Demonstrate functionality of DTN protocol for Earth-based applications using ION-DTN software

Model Logistics:

- **ION-DTN** software downloaded from SourceForge.net and loaded onto two Ubuntu 18.04 server machines

Model Validation: ION-DTN was developed by Dr. Scott Burleigh of NASA's Jet Propulsion Laboratory



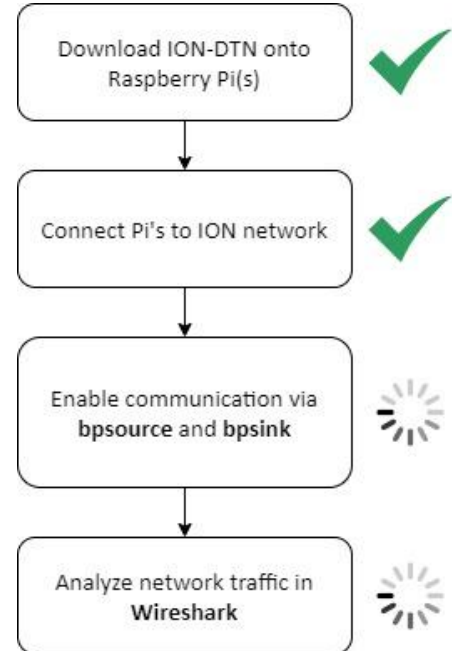
Hardware Demonstration Timeline

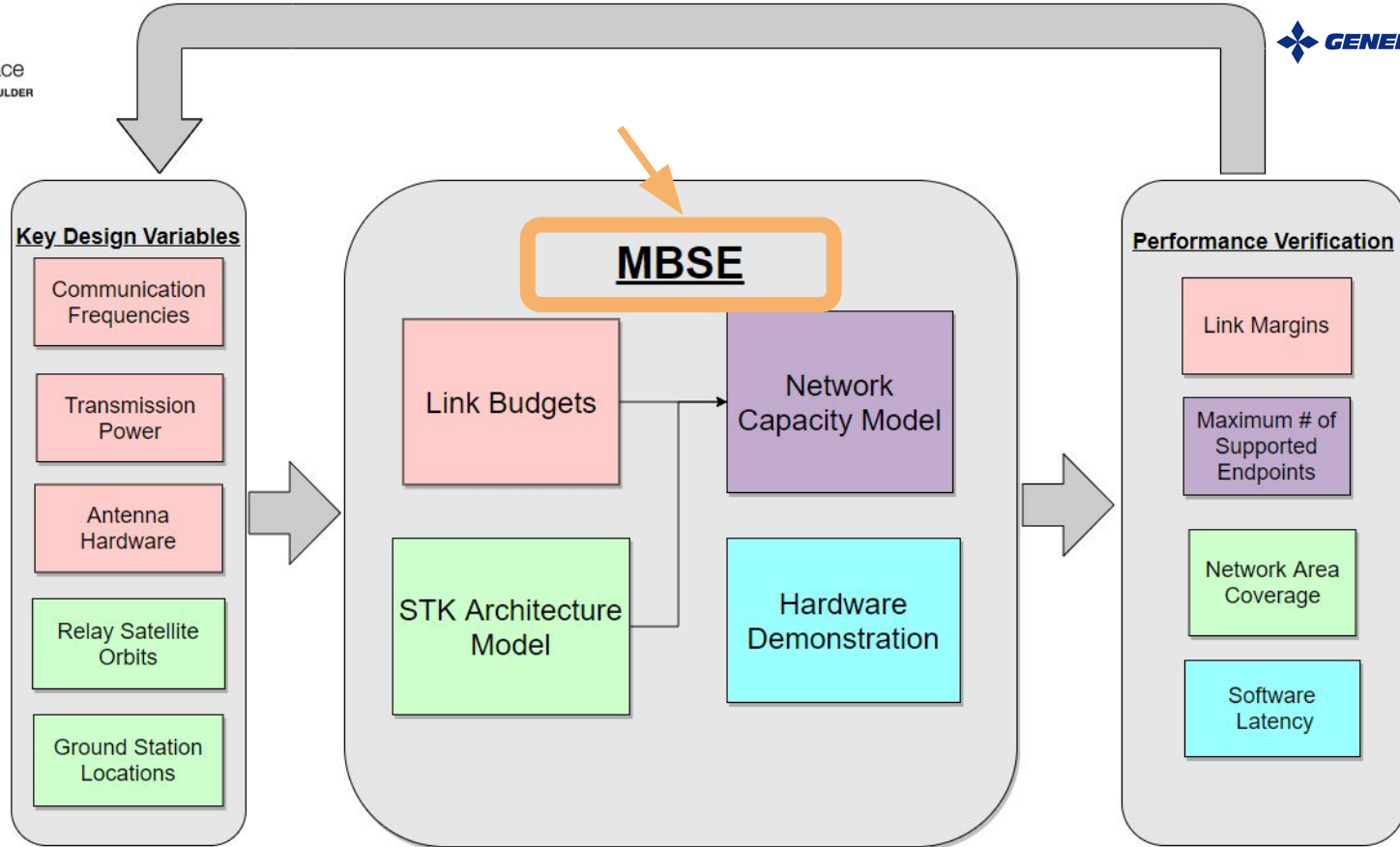
Current development:

- Ubuntu server & ION-DTN installed
- Proof-of-concept validated

Future Development:

- Enabling communications between two Pi machines
 - Must debug given **Assembly** test file
- Use **Wireshark** to calculate internal hardware latencies
 - **RISK:** time intensive, steep learning curve







SysML Modeling/MBSE



Purpose: Integration of separate project elements and model outputs (network capacity, link budget, etc.) into one project space, and trace requirements to the subsystems that satisfy them.

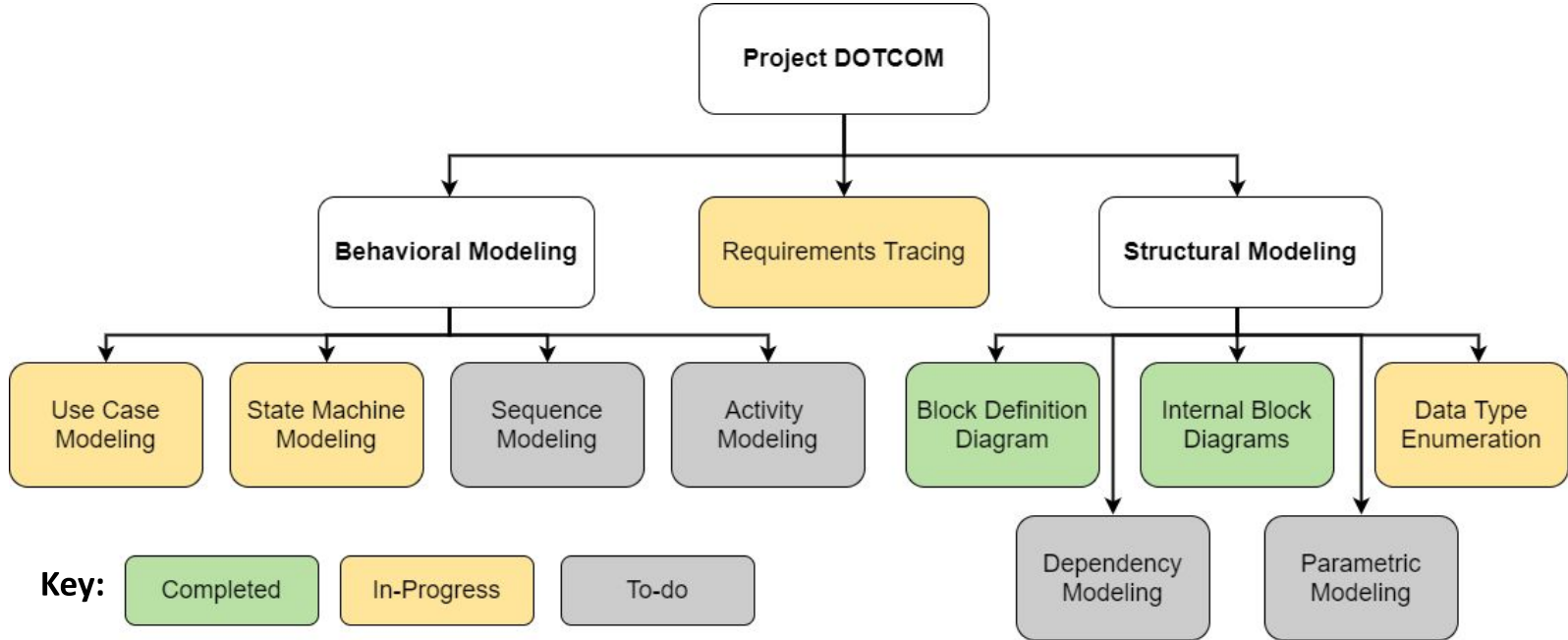
Model Logistics:

- Created in the SysML modeling language
- **Inputs:** Completed modeling of project subsystems
- **Outputs:** Cohesive DOTCOM project deliverable, including mapping of subsystem connections and modeling behavior of network nodes.

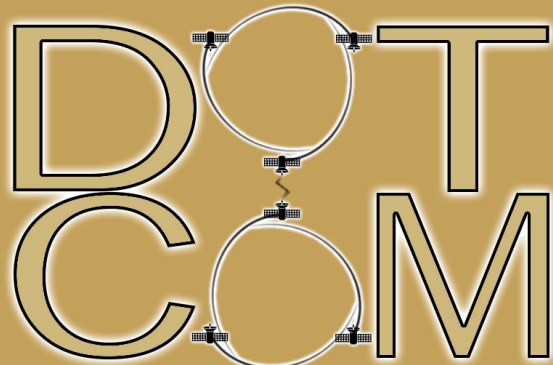
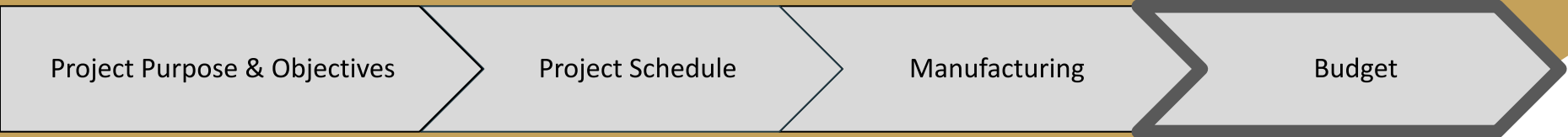
Model Validation: Validation of project inputs will come from their own verification and testing steps, as outputs from these models are loaded into the SysML simulation.



SysML Modeling Progress Update



Budget





Hardware Procurement Update



Hardware Parts	Status
2 Raspberry Pis	Received
2 Power Cords	Received
2 Monitors	Received
2 HDMI Cables	Received
2 Keyboards	Received
Total:	\$474.94

Hardware Parts	Status
5 Raspberry Pi	Yet to be ordered
5 Power Cords	Yet to be ordered
5 SD Cards	Yet to be ordered
5 HDMI Cables	Yet to be ordered
5 USB Cables	Yet to be ordered
Total:	\$984.40



Updated Budget

Key Hardware Components:

Raspberry Pi (7)

Monitors (7)

SD Cards (5)

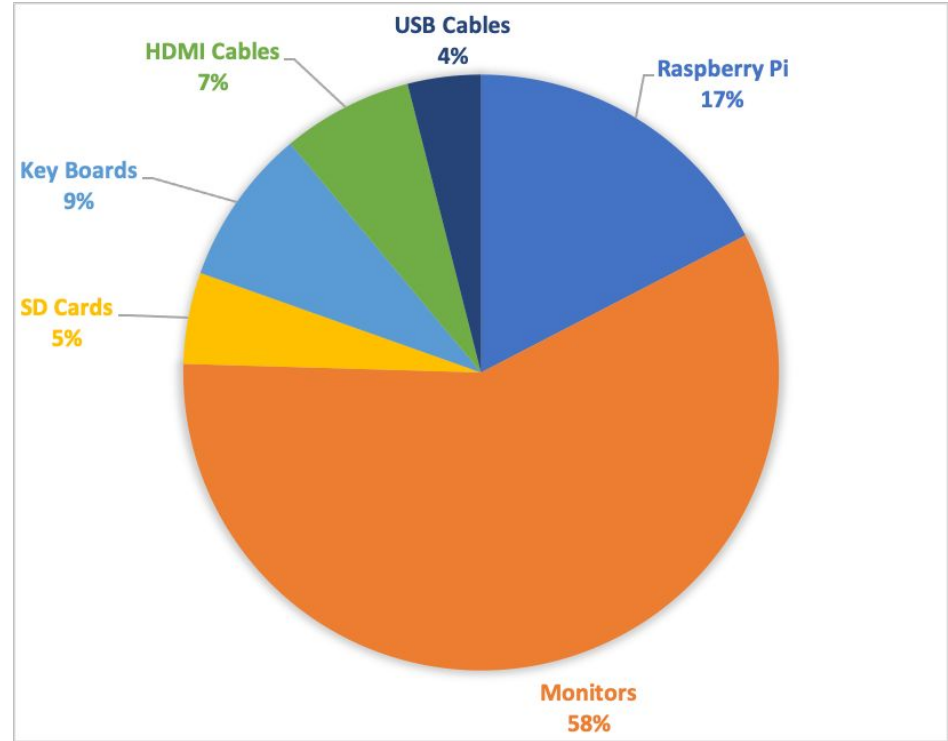
Keyboards (2)

HDMI Cables (7)

Power Cables (7)

Total: \$1,459.34

Margin: \$3,540.66





Smead Aerospace
UNIVERSITY OF COLORADO BOULDER

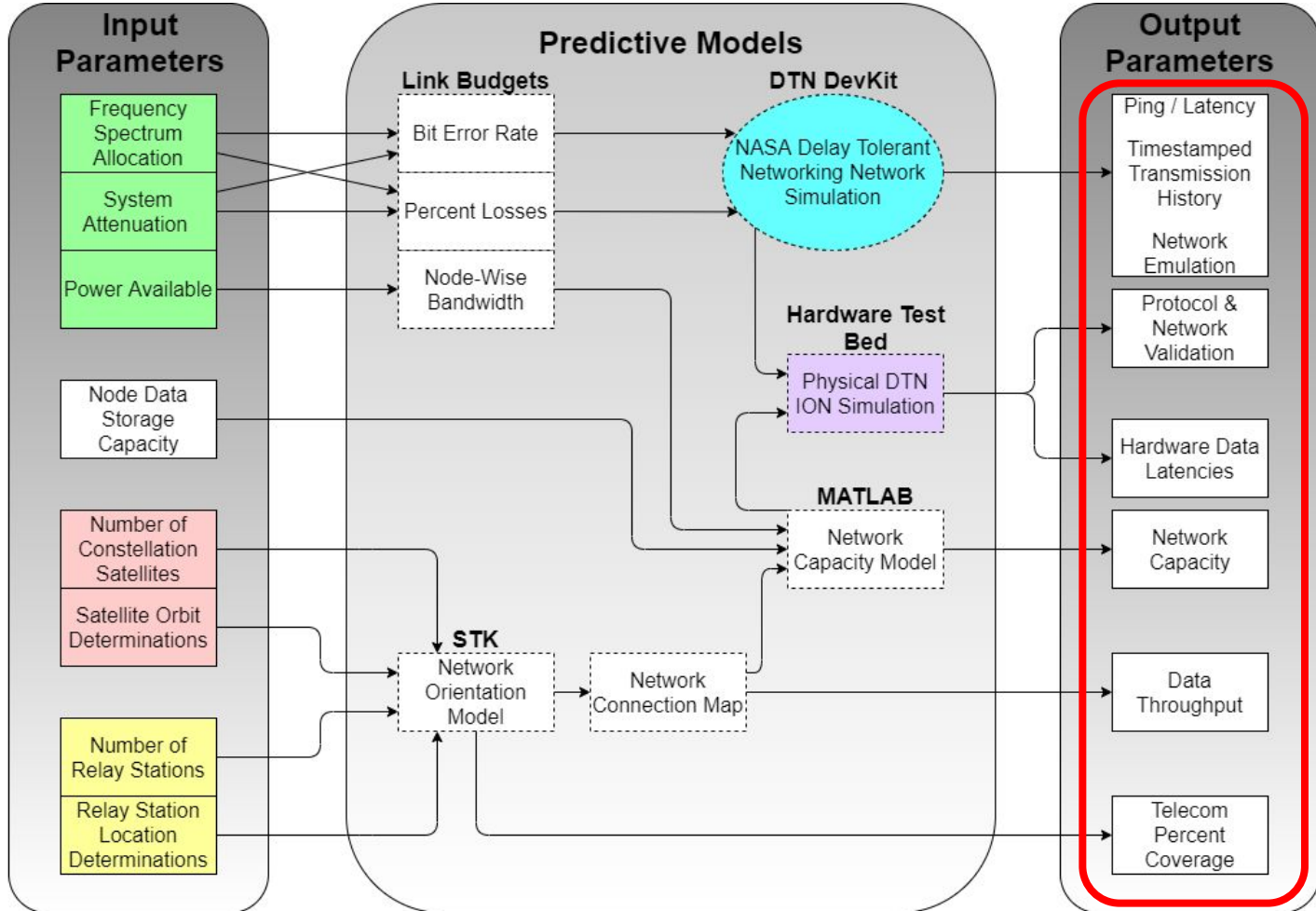
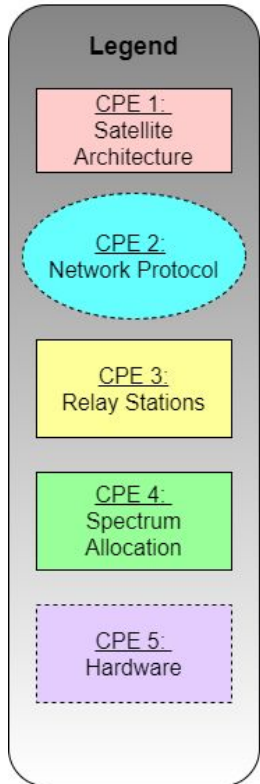


DOT COM

The text 'DOT COM' is rendered in large, gold-outlined, sans-serif letters. The 'O's are replaced by circular diagrams showing satellite orbits around Earth. The background features a vibrant space scene with Earth, the Moon, Mars, and a bright sun.

APPENDIX

System FBD



Relay Station Models

- Atmospheric Attenuation Evaluation (Matlab)
 - Input: Atmospheric humidity %, thickness of atmosphere, (elevation?)
 - Output: Signal Absorption (atmospheric loss variable)
 - Status: In progress
- Connection Window Evaluation (Orbital STK)
 - Input: Relay Station Positions
 - Output: Mapping of Earth relay stations' connectivity windows to the Moon relay station
 - Status: Not Started
- Model Verification (Orbital STK)
 - Repeat modeling techniques for well-documented system (Deep Space Network)
 - Compare modeled results to real-life behavior
 - Status: Not started

Constellation Architecture Testing and Verification

- Connection Window Evaluation (Matlab)
 - Input: Satellite position vs. time data
 - Output: Viability of each link at all times
 - Status: In progress
- Coverage Map (Matlab)
 - Input: Satellite position vs. time data
 - Output: Map of which points in the system have coverage at all times (target: >99%)
 - Status: In progress
- Model Verification
 - Repeat modeling techniques for well-documented system (GPS)
 - Compare modeled results to real-life behavior
 - Status: Not started

Constellation Configuration 6/6/4

Variables
Number of Satellites (T)
Number of Orbital Planes (P)
Satellite Spacing (F)
Inclination (i)
Right Ascension of Ascending Node (RAAN)
True Anomaly (v)

Configuration has format T/P/F

RAAN separation = $360^\circ/P = 60^\circ$

v separation = $F * \text{RAAN separation} = 240^\circ$

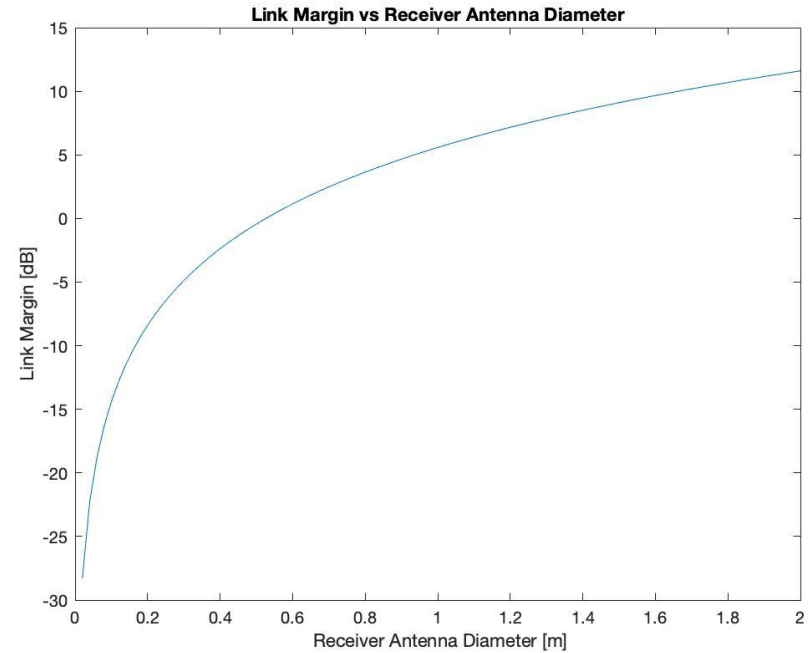
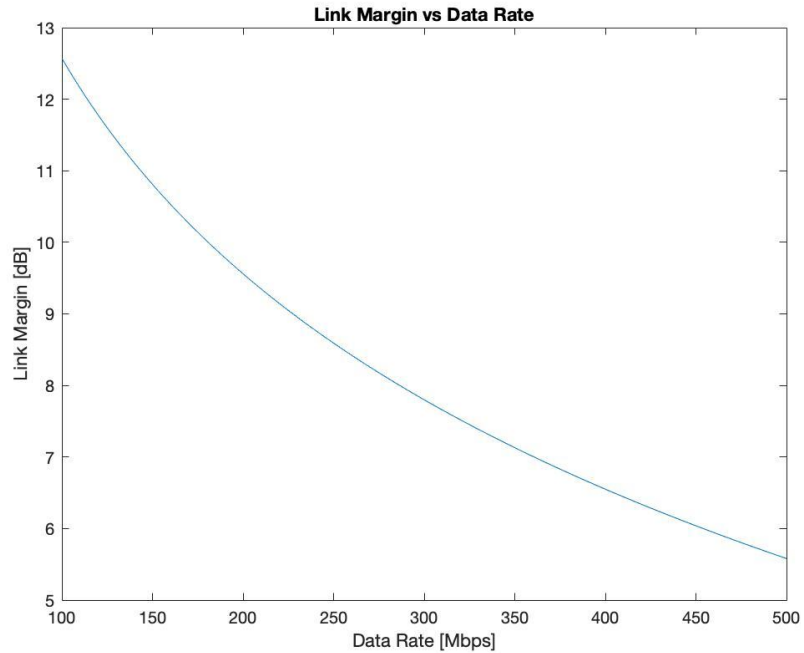
$i = 53.1^\circ = \text{constant}$

Baseline Parameters for Intersatellite Link

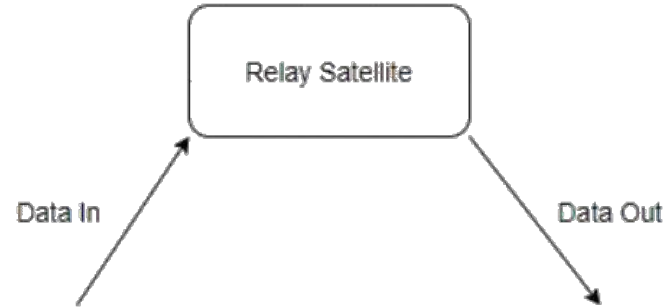
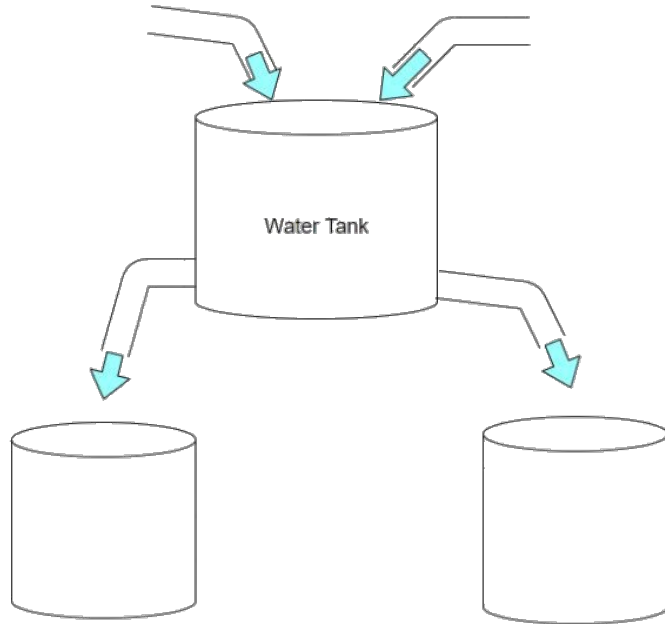
Key Input Variables	Value
Range (km)*	11,024 km
Frequency (GHz)	26
Antenna Size (D)	1 m
Transmit Power (W)	30 W
Data Rate (Mbps)*	500
Receive System Noise Temperature (K)	700 K
Required Eb/No (dB) [BPSK Modulation, BER = 10 ⁻⁷]*	11 dB
Required Design Margin*	3 dB

Outputs	Value
EIRP	61.9 dB
Antenna Gain	46.1 dB
Free Space Loss	200 dB
Received Power	-139.6 dB
Signal to Noise	19.5 dB
Link Margin	5.5 dB

Baseline Sensitivity Analysis Intersatellite Link



Network Capacity Model



Overflowing water tank is analogous to a saturated / max capacity network

$\text{Data Rate in} - \text{Data Rate out} = \text{Used Memory Change Rate}$

Network endpoints analogous to faucets adding water to the system



Data Rates



Forward Link Requirements

Data Type (Reliable Channel)

	Data Rates
Speech	10 kbps
Digital Channel	200 bps
Digital Channel	2 kbps



Element

Astronaut
Astronaut
Transport / Rover / Base

Data Type (High Rate Channel)

	Data Rates
Command Loads	100 kbps
CD-quality Audio	128 kbps
Video (TV, Videoconference)	1.5 Mbps

Element

Transport / Rover / Base
Astronaut
Astronaut

Return Link Requirements

Data Type (Reliable Channel)

	Data Rates
Speech	10 kbps
Engineering Data	2 kbps
Engineering Data	20 kbps
Video	100 kbps
Video	1.5 Mbps

Element

Astronaut
Astronaut
Transport / Rover / Base
Helmet Camera
Rover

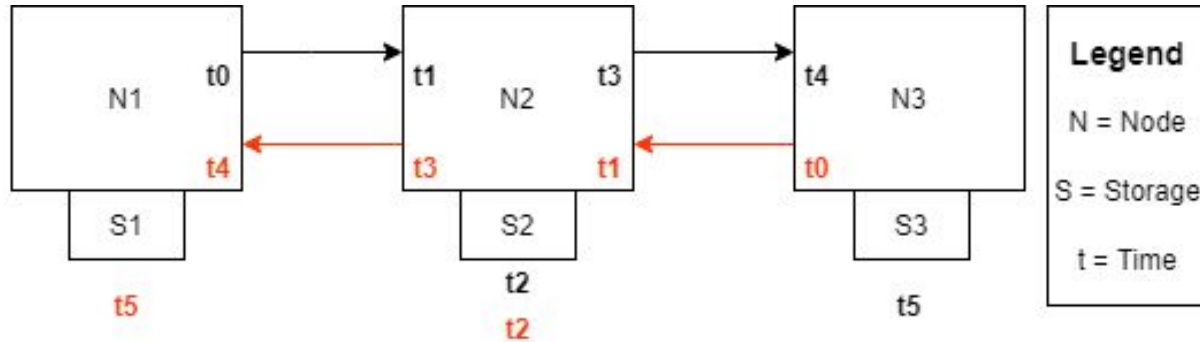
Data Type (High Rate Channel)

	Data Rates
High Definition TV	20 Mbps
Biomedics	35 Mbps
Hyperspectral Imaging	150 Mbps
Synthetic Aperture Radar	100 Mbps

Element

Astronaut
Astronaut
Science Payload
Science Payload

Key Measurement Methodology



Synchronous

$t1 - t0 = \text{Light Time Latency}$
 $t2 - t1 = \text{N2 Storage Latency}$
 $t3 - t2 = \text{N2 Forward Latency}$

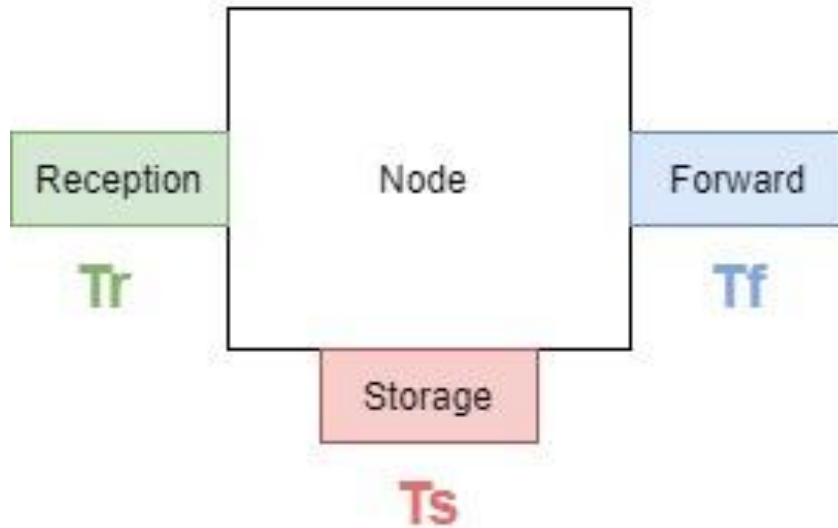
$t3 - t1 = \text{N2 Hardware Latency}$
 $t5 - t0 = \text{Total Signal Latency}$
 $(t5 - t0) - t4 - t1 = \text{Total Hardware Latency}$

Asynchronous

$t1 - t0 = \text{Light Time Latency}$
 $t2 - t1 = \text{N2 Storage Latency}$
 $t3 - t2 = \text{N2 Forward Latency}$

$t3 - t1 = \text{N2 Hardware Latency}$
 $t5 - t0 = \text{Total Signal Latency}$
 $(t5 - t0) - t4 - t1 = \text{Total Hardware Latency}$

$$T_f - T_r = \text{Hardware Latency}$$



Legend:

T_r = Time stamp
for reception

T_s = Time stamp
for storage

T_f = Time stamp
for forwarding

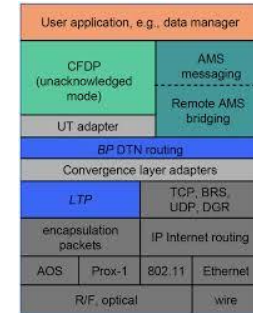
Completed testing

Completed:

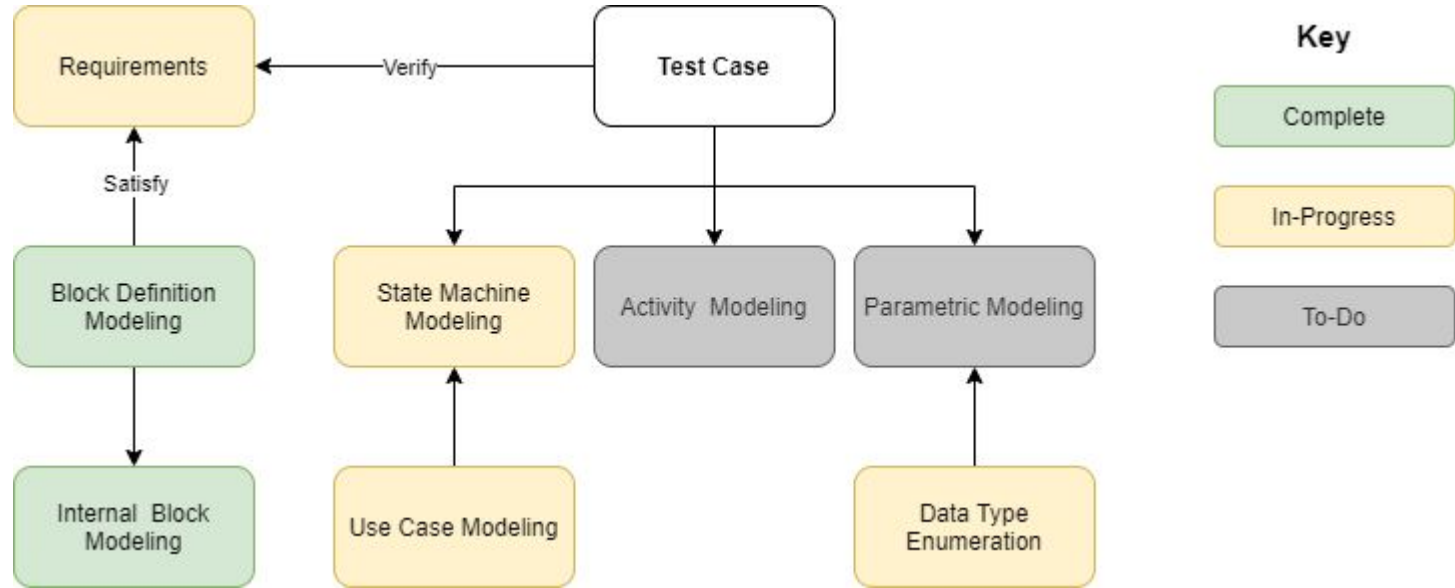
- 1) Ubuntu installed
- 2) Software installed
- 3) Software configured

In progress:

- 1) Debug given test file



SysML Modeling Validation



SysML workspace validation will be completed in part via test-case verification of other network modeling, as outputs from these models are simply loaded into the SysML simulation. SysML-specific verification is thus mainly redundant checking of requirements satisfaction/diagram completeness.



Current Budget

Key Hardware Components:

Raspberry Pi (2)

Monitors (2)

Keyboards (2)

HDMI Cables (2)

Power Cables (2)

Total: \$474.94

Margin: \$4,524.06

